

August 17, 1929

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AVIATION

The Oldest American Aeronautical Magazine



SPECIAL FEATURES

Final Preparations FOR THE AIR RACES

Expanding Markets WHILE PROMOTING SALES

AERONAUTICAL *Engineering* SECTION

**GROUND
HOLES**

As the various airplane engines continue to break endurance records, it is of interest to note that in every case Heald Grinding Machines have been used to finish the holes in nearly all the moving parts of the engines.

Some aviation companies, realizing that dependability and stamina of the airplane engine rests largely in the continual successful functioning of such parts as the cylinders, connecting rods, taper guides, gears, bushings, etc., are using some fifty odd Heald machines.

Having met the grinding problems of the automobile manufacturers for over twenty-five years, the Heald Service Engineers are in an exceptional position to offer suggestions on all ground and machine grinding jobs.

We urge that engines and parts be returned to them for complete details.

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The Heald Machine Co.
Worcester, Mass.



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Internal
Grinders**

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EAST ORANGE PLANT

of the

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*As a subsidiary unit to
be known as*

**ECLIPSE
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**ECLIPSE AVIATION CORPORATION
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DEFENDING THE
RECORD-BREAKING
ENDURANCE PLANE
The photograph shows how
planes subjected to the St.
Louis Robin were able to
be used by other pilots. The
planes were used to test the
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The Record Cracked *the Paint did not!*

420 hours, 21½ minutes of continuous flying! 420 hours, 21½ minutes of continuous strain on men, motor and paint! And, because none of the three cracked, the St. Louis Robin was able to add more than one week's time to the best previous mark for sustained flight.

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The reason, of course, is obvious. The fabric finish used on the St. Louis Robin was made by Cook. The dope used on the St. Louis Robin was made by Cook. All the varnish—all primer—all dope—paints used on this record-breaking

endurance plane were made by Cook.

The result was the toughest, most durable fabric finish possible to put on a plane. A fabric finish scientifically prepared to withstand the onslaughts of scorching heat, blinding rain, terrific winds—without cracking.

It is significant that only two of the last four planes to set new endurance marks reached the ground with the fabric in perfect condition—and that both of these planes were finished with Cook's Aviation Finishes.

We have some interesting facts about aviation paint for men who want the toughest, most durable fabric finish that money can buy. If you would like to have it, simply drop us a line. Your request involves no obligation whatever.



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S-W Nitrate Dopes made from latest solvents, give a superior

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Get expert information on combinations and new contrasts of two-to-light colors. Prominent members of the aircraft industry have been quick to adopt S-W perfected Aero Finishes which bear the hallmark of 60 years' experience and unsurpassed research facilities in the "finishing" business.

Write for the new booklet "Sherwin-Williams Aero Finishes" which includes a complete finish guide.

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Right: Series C "Wasp" cylinder and its 2 carburetors (upper member used in THE BATTLE OF BATTLES AIRCRAFT CO., Cleveland, Ohio). Below: Nickel Alloy Steel connecting rods and pistons. Right: Motor and fuel tank and fuel to Series C "Wasp" airplane engine.

Specialty material for the best customer of the world's greatest

Nickel Alloy Steel parts in the latest "WASP" aircraft engines establish new standards of wear-resistance

THE Pratt & Whitney Aircraft Co., in introducing the Series C "Wasp" engine, takes another step toward the perfection of the original "Wasp" engine brought out three years ago.

This latest engine retains all the fundamental features of design of the original "Wasp," but incorporates many new refinements which represent the experience gathered from many millions of miles of commercial flying and extensive naval and military use.

The Series C engine is stronger, more powerful, more durable, and generally more efficient than previous models, and its makers believe it is the nearest approach to mechanical perfection ever achieved in an aircraft engine.

Probably there is no better standard by which the engine can be judged than the results of the 50-hour endurance run required by Navy contract.

There were no failures of any nature during the entire test and the engine exceeded all requirements.

You are invited to visit our Booth at the 11th National Metal Exposition of the American Society for Steel Treating, Public Audubon, Cleveland, Ohio, September 9th to 13th inclusive, Space No. 94.

Nickel
FOR ALLOY STEEL

by a wide margin. It developed more horse-power, burned less fuel and oil, and, in addition, because of the confidence of its makers, was run under conditions far exceeding the set-down requirements. Throughout the entire 50-hour period, the engine was operated at full throttle.

At the conclusion of the test, the engine was disassembled and carefully inspected. Its condition was excellent. No parts showed signs of excessive wear. Of special interest is the fact that the big end bearing clearance of the master rod with the Nickel Alloy Steel crank pin was identically the same at the conclusion of the test as when first installed.

Probably the most impressive evidence of the uniformly dependable mechanical properties of Nickel Alloy Steel parts, is the fact that practically all manufacturers of airplane engines, both in America and Europe, have adopted Nickel Alloy Steel for highly stressed parts, the weight of which must be held to a minimum.



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A DELIGHT to the sportsman flyer . . . The Boeing 100, the civilian prototype of the famous Boeing army pursuit plane, claims the distinction of being the fastest sport model now on the market.

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Truscon furnishes you either the complete building from standardized units or the Steel Doors, Steel Windows and Stereolock Roofs adapted to your own design. Write us your requirements so we can offer suggestions without obligation to you.

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SIMPLICITY



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▲ ▲ ▲ ▲ It takes a specialist to produce a specialist ▲ ▲ ▲

THREE are three main factors to consider in the selection of a school of instruction in aviation. Are the courses offered complete and comprehensive? Is the flying equipment new and is it powered with modern motors? Are the instructors competent in every sense, experienced, and are they capable of developing a pilot's best talent? In other words, are the instructors specialists?

The vast difference today in the caliber of flight instructors results directly in the wide variance in efficiency of pilots who apply to the Department of Commerce for a license. And the same condition is responsible for the number of pilots the Department of Commerce refuses to license until these pilots have had additional instruction . . . at additional expense.

Spartan School executives believe the reputation of a school rests largely in the hands of its instructors and upon their ability to produce excellent pilots . . . specialists . . . without unnecessary expense to the student. Spartan instructors were chosen by that standard. They are outstanding pilots. They have both scientific knowledge and a background of wide experience. And they have character and personality which fit them to produce pilots who are specialists.

Behind this instruction staff is an organization which has supplied every detail of equipment required by a modern school of aeronautics . . . and financial support unflinching for the development of an outstanding institution, distinguished by the success of its graduate pilots.

Write for 32-page illustrated book . . .
"Training that Produces Natural Pilots"

SPARTAN SCHOOL OF AERONAUTICS
TULSA, OKLAHOMA

THANK YOU FOR SURVIVING AVIATION

What Facts "Stand Out" in Airport Lighting?

By E. J. DARLEY
Lighting Sales Manager
Graybar Electric Company

This is Number 4 of a series of advertisements whose purpose it is to reduce the volume on airport lighting to its simplest terms.

1. In general, what is the significance of good Airport Lighting?

The most important phase of commercial aviation today is the development of airports. Air traffic now waits on its terminals, and ninety per cent of all airline accidents are ground accidents. And of all ground accidents, as nearly every airport, the problem of night lighting stands foremost in immediate vital importance.

2. What is the commercial significance of good Airport Lighting? Does it pay?

The "edge" held by a first-rate airport is already assuming a tremendous commercial value. Important to mail and other concerns tend to seek the best equipped fields. What this may mean in the future is almost beyond computation.

The community possessing such an airport may expect generous dividends in the form of increased prestige, expanded business, and, more directly, the attraction of many new industries related to aviation.

3. What are the requirements for an "A" rating?

The Department of Commerce



See "What's New" in Airport Lighting



has seven requirements for an "A" rating: An airport beacon, Boundary Lights, Obstruction lights, Hanger

lighting, Field lighting, An illuminated wind direction indicator, A rotating beacon. And, also, continuous operation from dark to dawn.

4. What is the fundamental purpose of these requirements?

These requirements are intended to aid the pilot — to identify the field, to help him judge height and distance, and to show him the best approaches and the obstructions to be avoided; in fact, to give him a distinct, visible picture of the whole field.

5. In summary, what does all this accomplish?

A well lighted field must be easily found, easily identified, and as easy to land upon by night as by day.

And finally, it must be remembered that each field has its own individual requirements. The ultimate guarantee of good lighting is not equipment alone — but the proper application of that equipment.

Graybar's Airport Lighting Department — we may suggest at this point — stands always ready to offer its willing cooperation to those interested in this subject. Let this response bring you more information.

Office in 72 principal cities

Write for 32-page illustrated book . . .
"Training that Produces Natural Pilots"

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New York, N. Y.
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Graybar
Instruments or Station Electric Supply Dept.

AIRPORT • LIGHTING

THANK YOU FOR SURVIVING AVIATION

This speaks for itself

AMERICAN

CIRRUS MARK III



GREAT LAKES AIRCRAFT CORPORATION

CLEVELAND, OHIO

Mr. F. B. Everett
American Cirrus Engines, Inc.
Belleville, N. J.

July 4, 1939

Dear Mr. Everett:

Returned yesterday from Bradford, Pa. Based the AMERICAN Cirrus there Saturday. But best bit under the conditions don't think it was so hot. In the tests I found the limitation was air-cooled engine up to 500 horse power. I took fourth place against engine ranging from 118 to 148 horse power and so met some airplanes especially streamlined for racing. Better said next time.

Coming back from Bradford, I ran thru about four thousand feet. Pilot's loop for certain whether the engine would drop out but apparently water did not affect it. Then just before I reached Cleveland, I ran into one of the heaviest clouds to date but in the evening. The wind came down to about one third the speed and a wind very fast and violent. It had been raining and I had been in the air about two hours. Then in the storm I had to speed up to full throttle to hold my way, to descend all over the city, and landed in the field. But when I got to the water there was a 150 mph wind and when we began to leave I had 15 perfectly but it for nearly 30 hours the rain was so heavy the pilot of the engine was returned off the clouds before I saw the water and the engine was of the propeller was completely broken.

I am leaving tomorrow for Denver, where the AMERICAN Cirrus, will have some data on performance after the test conditions have been taken. I probably have several flying conditions on how to take country, on day, to have about twenty hours on the engine in flight, with perfect results. All temperature had some loss over 10 degrees Fahrenheit.

So far I'm actively satisfied and the more than I put on it, the more confidence I'm developing, particularly after the experience yesterday. Here later, as things develop.

With best regards, I am

Sincerely yours
Charles Mayne

AMERICAN CIRRUS ENGINES, INC.

WASHINGTON AVENUE

BELLEVILLE, N. J.

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CERTIFIED PERFORMANCE TRIALS AND SAFETY RATINGS

AERCO SAFETY RATINGS of aircraft are based on a thorough investigation of the aerodynamic characteristics of the aircraft, calculation of stresses in all flight and landing conditions supplemented by an investigation of fuselage strength in flight with the interference, selection and treatment of materials, dynamic and static tests of materials and assemblies, flight trials under operating conditions and other considerations affecting the safe and efficient performance of the aircraft.

Aerco safety ratings may also be awarded to power plants and accessories, after suitable investigation and tests.

AERCO CERTIFIED PERFORMANCE TRIALS are accurate and unbiased. Highly specialized test pilots and the finest equipment obtainable are employed. They are of inestimable value to the manufacturer who desires to meet competition with indisputable facts and to the operator who demands definite comparative knowledge of performance.

Regular inspection of production and of operating aircraft assures that the high standards of an Aerco rating are maintained. A high Aerco Safety Rating indicates that a product is the result of intelligent planning, honest work, expert craftsmanship, and real merit in aerodynamic design and construction, followed by proper upkeep in service. Aerco Ratings afford insurance and finance companies an impartial, accurate survey on which to base their rates.

The Aerco also maintains a complete aeronautical engineering consultation service, and Aerco consultants include many recognized leaders in the various specialized phases of that field.

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AVIATION INSURANCE PROTECTION

facts about B. & B. Service

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FIRST to supply, in 1922, comprehensive aviation insurance under an all-American combination policy form, thus which no broader form exists today.

FIRST to contract and supply aviation finance insurance, dealers' and manufacturers' blanket policies, airport and aircraft liability policies, and policies covering beyond the United States.

FOREMOST in constructing and supplying, at a moment's notice, special forms of coverage to suit unique requirements.

ALONE in operating on a non-pool basis maintaining an open market through which responsible insurance companies may operate independently.

ALONE in maintaining an engineering and advisory service covering the entire country and serving all interests.

ALONE in having developed a world-wide organization whereby American exporters of aircraft may secure adequate insurance and in addition, foreign local engineering and advisory service.

ALONE in having paid more aviation claims than all other aviation insurance organizations.

ALONE in being favored with the major part of all available aviation insurance business.

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"It's a great engine for sport planes"

POWER
SPEED
QUICK TAKE-OFF
FAST CLIMBING
COOL OPERATION
ECONOMICAL



From a dead start it fully develops its reserve power in a smoothly continuous climb into altitude with almost the ease of level cruising—what a thrill there is in flying with an Anson Engine! It's a great engine for a sport plane. A demonstration is the only way you can appreciate the exceptional performance of an Anson.

Literature presenting the improved Anson 2-cylinder model is available upon request.

Anson Aircraft Engine Co.
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Cotton Building, 101 and 103 E. 4th St., New York, N.Y.



ANSON
AIRPLANE
ENGINES

THANK YOU for maintaining AVIATION

An aërial view of SCHUMACHER FABRICS



Interior view of the "Aristocrat" a three-passenger monoplane which embodies the latest improvements and refinements in design.



Back of the seat cushions of the "Aristocrat" cabin is upholstered by the use of Schumacher fabric in shades of brown and beige combined by orange trim which matches the exterior of the plane.

YOUR own to command! A swift-winged plane with a cabin as luxuriously and tastefully decorated as a room in your home. The "Aristocrat" interior (illustrated here) shows how effectively Schumacher fabrics have been used throughout for decorating and upholstery.

The spirited modernistic design of the Schumacher brandings used on the back of the passenger seat introduces a smart contemporary note. Its undulating, zigzag stripes in graduated shades of brown against the color scheme of the ensemble which is brown and beige with an enlivening touch of orange. The seat is covered with rich brown moiré. An interesting new feature which adds to the refinement and finish of the plane is the windbreaker of durable dark brown silk and cotton creased the door.

The increasing use of decorative fabrics in airplane cabins gives fresh inspiration for Schumacher's modern weaves and patterns. In our collections you will find numerous and varied designs especially suitable for airplane interiors. Fabrics of smart simplicity and modern patterns and weaves in the bold spirit of contemporary design. Many of the thousands of private and passenger planes in use today are upholstered throughout in Schumacher fabrics.

F. Schumacher & Co., Dept. V-7, 60 West 40th St., New York. Importers, Manufacturers and Distributors to the trade only of Decorative Drapery and Upholstery Fabrics. Offices also in Boston, Chicago, Philadelphia, Los Angeles, San Francisco, Grand Rapids, Detroit.

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VILLAGE TOOL for marketing AVIATION

AVIATION

THE OLDEST AMERICAN AERONAUTICAL MAGAZINE

A MONTHLY-MEDIA PUBLICATION ESTABLISHED 1914

EDWARD P. WARNER, Editor

NUMBER 1... August 17, 1929... NUMBER 7



Air Mail "Tests"

TO THOSE who are professionally engaged in aviation, and who have as a matter of course kept in touch with each step in the evolution of America's air mail service, it seems almost incredible that any intelligent man can still be unimpressed on where that service goes and what it does. The hobbyist has been lost, long, and continues. The Post-office Department, the War mail contractors, the aviators and the general press, Chambers of Commerce and other civic organizations alike, and a host of private citizens interested in aeronautics or merely public spirited, have all joined in protesting from the test of the air mail's possibilities, exploring its present status, and telling how, when, and where it can be used to advantage. And yet, after all these years of effort, there come occasional evidences that the lesson has not yet been learned. The air mail service is still liable to be overlooked where it would be of value and should be used. Attempts are still made to use it where it can be of no service, leading only to disappointment and to criticism as sweeping as they are unjust. The work of instructing the public upon what they can really get from this new channel of communication must go on unrelentingly. It must even be intensified.

Only a few weeks ago the Washington Post, a leading journal in the nation's capital, conducted a test of the air mail. The results as the newspaper interpreted them were summarized in the headline, appearing across three columns of the first page, "Test by Post Reveals Air Mail Not Always Speedier Than Train." Subordinate heads, reinforced by a subheadline, explained that in 44 trials the airplane had beaten the railroad in 23, run a dead heat in fourteen, and been vanquished in seven. To the headline-skimmer and so dead to many of the Post's readers, it seemed a shocking record.

Shocking,—until the nature of the test is examined. After that it seems too strange.

The 44 trials, plus two uncompleted tests, corresponded to the 48 states. For some obscure reason the capital of

each state was selected as the starting-point of the run. If the air mail were a political service it would be appropriate to test its performance from political centers, but it is not. Its operation and its mission are not political but commercial. The state capitals, with hardly a dozen exceptions, are of insignificant importance as commercial centers, and they have properly been neglected in blocking out the first main lines of the air map. Only fifteen have a population exceeding 100,000. Only 20 of the 48 are touched by any air mail line, even the most minor feeder.

NO account was taken of distance. We have supposed for years that it was common knowledge among educated people that under present conditions the airplane is of little or no use over very short distances. We have entirely been wrong. In this "test" a race between "air mail" and rail mail from Annapolis, the capital of Maryland, to Washington, 35 miles distant by highway, was given as much weight as one extending across the continent. Annapolis not only has no air mail, it has no landing field. At least three of the "air mail" letters traveled all the way by train. At least three more were mailed from points within reach of an over-night run by train, and could not beat the railroad-borne letters until the newspaper office where delivery were to be made immediately after the arrival of the mail plane about two in the morning.

No account was taken of schedule. The air mail's operations have been meticulously planned to use a maximum of over-night flying and have a maximum of time out of the business day. Most business affairs, in spite of all plans to distribute the mailing load off the post, send out the great bulk of their correspondence late in the afternoon, and in most of the great trading centers the air mail means little to catch a without delay. Yet in this group of tests, nine of the 46 letters were mailed before eleven in the morning, 21 before one in the

afternoon, and eight at 7:30 in the evening or later. Hardly a fair selection of business result.

We have no desire to dwell upon criticism of the *Post's* scheme. Basically the idea was sound. It would be the last to claim that the air mail's reliability record has achieved perfection, or that further improvement is not needed, but the record being made even now is a good one. It is better than the general public suggests. It is certainly far better than the makers of the *Post* would have been led to suppose. The public are entitled to know just how good a service they are getting. Tests for fuel systems are to be honestly recommended—but they should be fair tests.

To revert to our introductory paragraph, that to mislead a group of readers should have been secured and circulated in good faith makes it evident that there is need for a vigorously continued educational campaign, with special reference to the latest mailing list in each eye to catch each phase. Air mail posters and stickers ought to carry that information in full whenever possible. For the largest cries, either the Post Office Department or the air mail contractor ought to prepare maps, better and more detailed than any that have so far existed, indicating the points of the country in which air mail can probably be used for mailings at various hours of the day. To withhold knowledge on such matters may drive in a few letters that would go by rail if the reader realized that the airplane could save him nothing at that particular time and on that particular route, and may slightly increase the revenues of some of the contractors, but in the long run it will be most profitable to have given the public all the facts all the time. A single dissatisfied customer can give the air mail service much more than five cents worth of undesirable publicity.

//

Gate-Cramming Epidemic on Aircraft

IT ALL BEGAN with Clarence Torbush, a youth whose biography, as patently indicated by the gynaecologic press, indicated him to have a fixity of purpose in life which is unhappily met among young men. Clarence, if his past history was accurately reported by the newspapers, had chosen the career of going on trips where he was not invited and was not wanted, and had pursued it with a success that attained a climax when he crept on board the Graf Zeppelin just before her departure from Lakehurst for Friedrichshafen. Sob stories unbesieged their typewriters to extol him as the embodiment of the spirit of youth. His fellow-countrymen who wrote letters to the press to make public display of their delight over the success of the east trip that he had played had the satisfaction of seeing themselves outdone by a part of the population of Friedrichshafen, who

fairly took Clarence to their hearts. Six months ago the "use and only original Zeppelin story" was still a valuable headline, but we have enough faith in the collective good sense of the American people to trust that they have turned their backs on him by now.

Clarence may still be original, but he is no longer the use and only. After his crew Arthur Schreiner, and the just began to fail. It needed neither a pilot nor an aeronautical engineer to perceive the glaring redundancy of arbitrarily adding a hundred and fifty pounds of useless weight to a machine already loaded to the very limit and balanced with the last degree of delicacy for a take-off. Arthur found himself lost of a hero that he may have anticipated, and at last accounts his story of his adventures was still unpublished and his theatrical contracts were hanging fire.

NO ATLANTIC FLIGHT was being complete without its showman. Albert Bandier thoughtfully added himself to the complement of the Graf Zeppelin when she left Friedrichshafen two weeks ago, and thereby gave Doctor Hugo Eckener a chance to establish for himself a new fame as the man who solved the showman problem.

To have done the obvious thing, and to have passed to lead the showman passenger at once would surely have insured that several readers would have had their place further in advance and considered themselves more carefully when the next trip was made. To look him up in a freight compartment, refusing to discuss him with the passengers or even to give out his name, and to turn him over to the American authorities for immediate deportation, took all the romance out of his performance. That any scene in the aftermath of previous showmen and their exploits to have been cruel and inhuman treatment. Admiration of cold common sense, and believers in aircraft as other instruments of mass transportation enterprises, will find Doctor Eckener's course with a right of mind, sustained only by regret that our own metropolitan officials felt it necessary to provide the opportunity of even a little increased publicity for the "young man of personal ideas" by allowing the press photographers and interviewers to have access to him.

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Thirsting for a Thrill

WE HAVE BEEN SLOW to criticize the daily press for their reports of aerial crashes. The reports are often inaccurate, and they are often played up with an excess of sensationalism, but we give credit in most cases for an honest attempt to state correctly facts that are often difficult for even an expert to determine. We have no desire to join in the search-baited sometimes heard for secrecy, or for those

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AVIATION August 17, 1929

ting the stories of aerial mishaps down to below what their real importance, justice, or significance should entirely. Only by knowing the truth about what has gone wrong can we present a reasonable view of the trouble. Only by dealing with the public with full candor can we prevent the spread of rumors sure to be much worse than the facts. Even now, the average newspaper reader has an exaggerated idea of the dangers of flight because of the misapprehension of the information being open that must that he is actually receiving. He fears frequent reports of accidents but he has no grasp for himself, unless he follows very carefully the releases of the Department of Commerce, at the total number of mishaps and the total amount of flying done. This gains is commonly worse than the reality.

We do, however, feel entitled to protest when the desire to find a fresh excitement for each edition leads to the broadcasting of untrue facts of impending dangers of which it is never the slightest probability. To throw the accident countesses into the shadow of almost headlines is only one degree less objectionable than deliberate falsification, and it is for those reasons.

THIS LAST ZEPPFLINE is on her way back to Germany at this time. The photographs are being shown. The coming will automatically repeat the experience of the preceding three. The headlines in a certain type of paper will scream: "Airship thought to be at danger", "rolling head first into the sea", "Zeppelin approaching giant storm area", and the like. Even if the statements be literally true, they are outrageously misleading—and the editors must know it. Some of them might with equal accuracy have been applied to the Bremen during her recent maiden voyage, and with the same implications, were it not that the general public is too well versed in the performance of ocean liners to accept such hints concerning them at their face value.

There is no way of controlling the newspapers that elect to sound the sensational note. If misleading headlines will sell papers, some journals will continue to use them. We can, however, make ourselves heard in our own communities, and we may be able to convince some publishers among that group that to treat every novel or notable flight, whether of airplane or airship, as a prospective tragedy, is unprofitable practice. Papers that do that may gain some temporary profit, but they will alienate sober readers. In the meantime, they deserve not a penny's support from the aircraft industry. Every aeronautical advertiser, in his dealings with the general public, has a responsibility towards the industry of which his company forms a part. The periodical that makes a conscientious effort to give a fair and complete rendering of aeronautical news deserves the backing of aeronautical people. In the long run, too, the paper that shows inaccuracies, and misleading sensationalism will prove the most profitable medium.

What Constitutes "Refueling"?

WHEN CAPTAIN LOWELL SMITH, late of the *Enterprise*, was shot down the *Enterprise*, and Lieutenant Richter did his own plan of refueling the airplane from another in flight, the international refueling bodies were virtually to deal with the problem presented. The scheme was so novel that there was nothing in the rules governing aviation flight to prohibit the taking on of supplies elsewhere than from the ground. The International Aeronautical Federation was prompt to deal with the emergency, and to put refueling records in a separate class.

Those were simple days. No one even then thought of supplying the record-making plane with anything but fuel and oil, and possibly a few simple supplies for the pilot. We had progressed, and the facts now involve specially prepared hangars, refueling aircars, solar stills, window shades for the cabin, and parts for the engine. At this point an interesting new problem arose.

Officer and Jackson during their recent successful attempt on the record were thus once accomplished the remarkable feat of changing spark plugs in flight. That they did so, of course, in no way way impairs the significance of their flight. The plugs are light accessories which are known to be apt to give trouble after long operation, and an ample supply for the whole flight could easily have been carried in taking off. To change plugs, in other words, is quite as preferable as a non-refueling as is a refueling trip. But where do we go next?

FOR A MULTITUDE OF PLANE on which one engine can be stopped without interrupting the flight, it is but a step from the supply of spark plugs to the provision of valve springs, valves, or even cylinders and other parts of larger bulk. It is not even inconceivable that a complete new engine of moderate power might be received and installed in flight. With those possibilities in prospect some very interesting records could still be made, but they would not be the same type of record that has been established so far. The interest of refueling flights as a means of engine testing would have been diminished if not destroyed.

It has been plentifully demonstrated that the continuous running life of a good modern engine without any replacement is as long as anyone could reasonably wish that a plane should stay aloft. The possibilities broad at hand should be not before they arise. The National Aeronautic Association, as the official body controlling aeronautical sport in the country where refueling flights have been most popular, should take the lead in presenting to the Fédération Aéronautique Internationale, the well-known F.A.I., the recommendation that no transfers of engine or airplane parts from the refueling plane should be permitted.

of last year's races, which were shown before dozens of clubs and civic organizations of the city. In the meantime, I was also interviewing applicants for positions in the organization. This was arduous work because of the glare we look in selecting our personnel. We have made it a policy not to let anyone having a background of preferential promotion work. One reason for this is that our project is designed as a constructive merchandising medium, and not as an entertainment project. The entertainment features are purely incidental. If the National Air Races feel to broaden the aircraft market they should be out on the boards. The Cleveland races are also designed to do so in the truest sense a national event, being so geographically every section of the country.

This year we will introduce a number of innovations on the National Air Race program, some of them so daring as to be radical. The worst air derby from Los Angeles to Cleveland is one such event. It is the second class's first real pressure signifying the entrance of "wildcat" into aviation. Such a departure is attended by a lot of responsibility, but the participants of women in flying activities demanded it. The races would not have been complete without it.

"Another, and one of the first innovations designed in the program, was the all-Ohio air derby. This event in which the contestants will fly around the state of Ohio, was created to give the state's background of local competition. The introduction of gliders and the staging of the first national glider competition should give an impetus to great reduced expenses in glider activities. The entrance of lighter-than-air craft is another important progressive feature.

"This year, also, we will have 900 estimated and confirmed spectators moving about in a decorative scheme compatible with the race track and exclusive banquets. Every part of the ten days' activity, in fact, has been designed with this idea in view. Some of the decorative scheme and service facilities to be enjoyed this year are a further development of ideas originally introduced at Los Angeles last year. Many of them, however, will be seen at the air races for the first time."

One of the most radical departures from our race program to be seen at Cleveland is the introduction of "home race starts." It involves starting all of the contestants in a given race at the same time, after which the participants fly a "scattering lay" before beginning the laps around the smaller triangular course. A "scattering lay" is located 30 miles from the airport. The in-

tercourse must circle this pylon and return to the field before starting the great around the smaller course. This feature was introduced to decrease hazard and to make it easier for spectators to know how the race stands at any stage of the contest. Mr. Henderson expects the "scattering lay" to revolutionize closed course airplane racing.

THE STAGING of an international exposition along with the national air races was introduced by Mr. Henderson at Los Angeles last year. Although the exposition attracted large crowds, it was found that the housing of such a show on, or too near, the airport is impractical, since the patrons are lured to the open field by the heat and noise of planes in action. This year the exposition is being held concurrently, but in far enough away to be non-competitive. The exposition is designed as a retail selling show, and undoubtedly will attract many pilots. Unlike all previous aircraft shows, the Cleveland exposition will have certain hours daily set aside for the industry, during which the public will not be admitted.

The problem of having a properly equipped airport—a most essential requirement in a project of this kind—has been met by the city of Cleveland, which is spending \$450,000 in modernizing the municipal airport for the races. Like the city government, the county officials have also realized the magnitude of the event and are equipping and widening eight miles of highway to facilitate the flow of traffic to and from the airport.

An interesting difficulty confronting Mr. Henderson and his aides was the necessity for leasing property adjacent to the airport in order to prevent a shortage in gas receipts. Past experience has revealed that hundreds of persons who can find a vantage point on property adjacent to the airport property will not pay in the regular way to witness the activities on the flying field. That this condition might be eliminated in Cleveland, the executive committee decided to lease as much of the surrounding property as was thought advisable.

Since practically all of the property sought was subdivided and the owners scattered all about the United States, one can imagine the work the leasing process involved. However, the owners were located and 80 per cent of the required property, approximately 1,800 acres, leased at \$1 per parcel for the period of the air races. Through the co-operation of the Ohio National Guard, a force of 200 national guardsmen will guard the leased territory, keeping it clear of persons who do not wish to pay the \$1 admission fee at the gates of the airport.

Expanding Markets WHILE PROMOTING SALES

By JUSTIN A. MCINANEY

First President in Charge of Sales
Alexander Aircraft Company

BY FAR the greatest strides forward in commercial aircraft manufacturing today are being made in distribution. Plans and production methods are essentially the same as they were two years ago although material improvement has also been made in this direction.

Stimulation in aircraft sales methods may be traced to the introduction of cheap war production coupons, which for a time placed manufacturers of the popular three-place open plane in a doubtful situation. Should they attempt to maintain heavy production schedules, knowing that the planes, powered with expensive new-producing air-cooled engines, would sell at retail far double their price with C-X-S's?

Many observers predicted a slump in aircraft production. This has failed to materialize. Why? The reason lies chiefly with better sales promotion on the part of the more progressive manufacturers.

In the Alexander Aircraft Company, for instance, we have enjoyed particularly good success with new plans for selling on terms payments, and for creating active flying sales promoters.

The sales promotion plan has been received with some skepticism by the industry and mistaken alarm on the part of one or two of our distributors, who have given the impression that we intend to use them with their consequent territorial protection. Naturally, we would not consider such a step unless, in our franchise specifications, the territory is not being properly worked. Our sales promotion efforts are confined almost entirely to rich open territories.

Briefly stated, we are leasing transport pilots a leased new Eaglehawk demonstrator with which to tour open territories, make sales on fifteen per cent commission, keep passengers to delay expenses, and make money both for themselves and the factory. Two transport pilots have already taken out demonstrators and are in the field. The first to do so was Joe Ben Lieve, senior war nurse instructor and test flyer, now a commercial operator with headquarters at San Antonio.

The industry certainly cannot quite realize the idea that we are willing to own over a new airplane to a pilot for him to use for personal profit. Naturally, we prefer



Justin A. McInaney

caution from increased sales and greater popularity of the plane.

The sales promoter pays his own operating expenses, which may be defrayed by barnstorming, and of course is responsible for damage to the ship, reasonable wear excepted.

A fifteen per cent commission is allowed on each plane sold. If the sales promoter buys his demonstrator, we allow a twenty per cent commission. The salesman can pay gas and oil expenses by buying passengers. In exchange for the use of the plane in passenger work, the factory makes a small hourly charge, usually around \$1, to cover depreciation on the engine.

We were led to try the plan by a desire for more active representation in open territories. The more representatives the more sales. This being has demonstrated its truth in the experience of the Alexander Film Company, our sister organization.

THE PLAN enables an automobile salesman, for instance, to capitalize on a desire to learn to fly and enter the gate on a business basis. He can hire a transport pilot on a split commission agreement, take out a new Eaglehawk and leave to fly on the road while earning revenue from his sales and passenger receipts. Our sales force includes eight distributors and dealers who are also selling automobiles, and there are about 25 Eaglehawk distributors and dealers who were formerly in the automobile industry.

Automobile men usually make good airplane salesmen. They have the sales experience which many pilots must acquire. On the other hand, persons unfamiliar with the industry must learn the fundamentals of flying so as to present an intelligent sales talk. Recognizing these facts, the Alexander company opened in July one of the first



Crowds flock to automobile operators at the National Air Races

Sheddenhouse estimates that a bearing surface of only 20 per cent of the total area is all that can possibly be obtained by the scraping method, while the accuracy is increased to from 92 to 95 per cent by line honing. As a consequence, the main bearings are line honed with fly-cutters after they have been lightened. In this, a check is made with the mesh of the honing gears and the depth of the thrust bearing, so that the gear will have the proper clearance and the thrust bearing will not be too tight in either half of the crankcase.

THE FLY-CUTTERS, of course, will bore bearings of varying dimensions and in the OX engines they are bored to allow a clearance of .0015 in. It is said that with this clearance a perfect film of oil is maintained over the entire bearing surface at all times. The clearance is not large enough to cause the oil pressure to drop below normal, and yet it is not so small that it causes the bearings to heat up and warp. With the .0015 in. clearance on the main bearings, an OX crankshaft should spin around three times when given an arbitrary blow with a fist. This test is used at the Parks School, although the fit is always checked by the application of Pennwalt lube. It is interesting to note that the war-time production thrust bearings are returned to overhauling the Parks engine. However, all bearings of this type are first sent to the Allibert Bearing Company in St. Louis for reconditioning. This firm repairs the ball races and the oversize balls are then, Mr. Sheddenhouse claims, outsize oversize over the work performed by the St. Louis firm, and says that the returned bearings are much better than new ones.

The next step in the overhaul of the OX engine is the checking of the crankshaft bores in the crankcase. If they are found to be in a roughened condition, they are bored with fly-cutters to bring .005 to .030 in. oversize. If the crankshaft itself is rough, it is ground just as was the crankshaft. The crankshaft bearings, which were returned in the first article, are mounted in an alloy of 115 brass test hardness. These are machined to be sure as to allow a clearance on the shaft of .005 in. In assembling the crankshaft, equal space is allowed on each side between the cam lobes and followers. The

position of the shaft is regulated from the forward cam bearing, allowing a clearance of .0025 in. between. After installing the crankshaft the crankshaft is placed in position. Oil is applied to the main bearings, and they are tightened. The timing gears are then added, and are checked so that their depth is absolutely correct, allowing for .003 in. back lath on the timing gears and from .012 to .015 in. on the oil pump drive gear.

The connecting rod bearings are fitted with the case crimp and the clearance allowed for the main bearings. In this work a Stern connecting rod-bearing machine and a connecting and bearing tool attachment for the South Bend electric drive belt lathe, which is included in the equipment of the overhaul shop, have proved invaluable. Through the use of these machines the connecting rod bearings may be bored either with the gear in place or removed. This is important from the standpoint of saving time.

THE PARKS ORGANIZATION has a very large number of OX-3 engines on hand, which are to be overhauled as quickly as possible, so that they will be ready to spare for the training phase, so that they may be installed in the Parks open cockpit P-41 biplane, which will soon be placed in production by Parks Aircraft, Inc., and so that the Parks organization may coordinate an overhaul and exchange service on OX-3 engines, which will be discarded later. It has been found with testing of these engines that the bearings are worn, but the pistons and the piston pins are practically new. In cases of this sort, the Stern boring machine is used, since this device will bore



Above—The Stern connecting and bearing machine. The jig provides boring the bearing bore with reference to the piston pin. Left—The main bore of the crank. Right—The main bore of the crank. Right—The main bore of the crank.

a connecting rod bearing without accelerating the removal of the piston. With this machine, the connecting rod and piston are held in perfect alignment in a jig, while the bearing is bored true with reference to the piston shaft. If for one reason or another, the piston must be detached from the connecting rod, the bearing is bored with the attachment for the lathe.

Overhaul pistons are not utilized in the engine now handled at Parks Air College. In case a piston is worn to such an extent that it cannot be used, or if it is broken, a new piston made by McGuyre-Norris to the old Government specifications is substituted. The same thing is true of cylinders. If there is a weakness on the part of a piston in which a cylinder may be cut very lightly, but not sufficiently to be around a springing operation. If a cylinder is in such a condition that it needs regrading, some part of it is usually found to be broken or cracked, so that it must be replaced anyway.

In a case where a piston is serviceable, but the piston pin bosses are worn, the bosses are turned out vertically to "clean up." The turning is done on a Van Norman reamer drive, making use of a Vatech drill with an aligning sleeve. In such instances, aerometric piston pins are employed. After the pistons and the cylinders, these are considered in the same way as the oil cooler than the new specifications and are made from .002 to .003 in. larger in diameter than normal, depending upon the requirements. In assembling the piston and the connecting rod, the piston end of the rod is turned with a blow and a pin and the pin is inserted through the piston and the rod. The hole for the lock screw is located at the same time by means of a special tool developed by the Parks organization.

It has been found that the application of heat to the assembly operation is much more satisfactory than running out the gas cylinder hole in the connecting rod screw.

If the latter method were employed, and the lock screw were to break, the pin might slip out of position, causing damage to the cylinder wall. The oversize pin used in overhauling the OX engines are safe, so that the application of heat does not cause them to temper down, or become too hard. A hard pin probably would crystallize and break when brought in contact with the hot end of the connecting rod. The soft pins, incidentally, are polished first in the piston at room temperature, providing a means of good lubrication when the engine is in operation. The bearing surface cleaned by running the piston houses is said to be about 80 per cent of the total area.

AFTER THE PISTON and the rod have been assembled, the assembly is placed in a Stern connecting rod aligner, which fits on the stand used to hold the Stern connecting rod boring tool. At any time the rod aligner corrects any twist or bend in the rod, aligning the rod to the full length of the piston skirt, and thus eliminating the possibility of alignment in a warped position. The machine has been shown to be accurate to .0025 in. in tests made at the Parks shop.

The next job is that of fitting the piston rings. Standard Ford compression rings are used, but so of standard rings are fitted. Mr. Sheddenhouse points out that so long as the connecting rod bearings are fitted properly, there is no need for oil rings. He says that they reduce the lubrication of the cylinder walls, and for that reason cut down the number of revolutions per minute that it is possible to obtain from an engine. They also cause an engine to "spit" because of insufficient lubrication.

something with which the Parks organization is never satisfied. Following the fitting of the rings, the re-setting rod machine is attached to the crankshaft and the test, using the pressure tank and the two gaskets of light oil, is again conducted. If any leaks show themselves, the engine is at once disassembled and the faulty work is corrected.

WHILE THIS PART of the work is in progress, the cylinders are also undergoing overhaul. Of course, they are first examined to determine their fitness for further use. The next step is that of inspecting the valve guides. If worn too much, they are replaced with new Miller guides, otherwise the old guides are reamed and that they measure .01 in. oversize. In the reaming operation, a Miller jig is used. This tool, which is said to be extremely accurate, is also used in both reducing the old and in installing new valve stems, the same operation, so far as the cylinder is concerned.

In reaming the seat, the jig is attached so that it is aligned with either the intake or exhaust port in the cast away bar, and with the support flange at the top of the cylinder, holding the reamer in the proper position in relation to the seat. So there is no danger of cutting into the water jacket, the reamer is run in a controlled manner until the port is level and threaded, and a new seat is put in. The joint, in a case of this sort, is sealed with a mixture of kerosene and glycerine, and the new seat is tested up. Incidentally, the water jackets are tested for leakage before the cylinder is given a final "O.K."

The old valves are used in engines undergoing overhaul if they are found to be in a serviceable condition. They are tested up on the South Bend lathe through the use of a tool post gage in conjunction with a special valve check. If the old valves cannot be tested up by this means, they are replaced with new Inland valves, which are fabricated of a metal that will not take the cast iron seat of the OX-3 engine. The process of grinding in the valves, the next step, is said to be of short duration as a result of the accuracy with which the valves themselves are tested up and the accuracy with which the work on the valve seats is done.

Like other parts of the engine, the valve actions are completely disassembled in the overhaul job. The top pin bolts are removed out of position, so that larger pins on the intake and exhaust stems may be installed. This is to eliminate any side wear on the valve guides. The hose connections and clamps are always replaced, and the intake and rod springs and straps are tested and checked for wear. The valve springs, by the way, are always replaced. When this work has been completed, the cylinders are assembled on with clean oil and are loaded in the crankcase, two paper gaskets that have been soaked in oil being used to seal the joints.

After bolting the cylinders to the crankcase, the gas-bore intake manifold is made up, so that there is no possibility of its cracking when it is stretched. The water manifold and the valve stems are also made up. The assembly of the water manifold is interesting, since the gasket joints at the water outlets are the only ones on the engines overhauled at the Parks school, which are sealed with shellac. Two gaskets are used to take up the expansion, as they are the last of the cylinders. Next, the valve dimensions are adjusted to .001 in., and the engine is fired by adjusting the main gear.

The third and concluding article of this series by Mr. Weber will appear in an early issue of AVIATION.



THE RYAN B-5 *Brougham*

EMBODYING a number of refinements in design and equipment over the old B-1 and B-3 Models, the new Ryan B-5 Brougham is now in production at the Mohawk-Ryan Aircraft Corporation factory, Anglen, St. Louis County, Mo. One of these planes already has been used in a record flight made by Miss Marnet Croase in California. The Ryan B-5 Brougham is powered with the Wright New Whirlwind 300 hp engine, which gives it a high speed of 140 m.p.h., a cruising speed of

120 m.p.h. and a service ceiling of 16,000 ft. In test flights the plane has risen off in 275 ft. and shown a climbing ability of 1,300 ft. per min. The landing speed is 50 m.p.h.

As to the ease of its performance, the new Ryan Brougham is a general purpose plane suitable for civilian air transport functions, private ownership, aerial survey, scout, instruction and other services. The standard finish is black-pine with flannel-lined padded seats with red trimmings. Special colors can be provided.

The Ryan B-5 Brougham has a wing span of 42 ft. 4 in., a length of 38 ft. 3 in. and an overall height of 9 ft. 9 in. The weight empty is 2,250 lb., the useful load 1,735 lb., the payload 1,000 lb. and the gross weight loaded is 4,000 lb. The airplane is manufactured under Approved Type Certificate No. 143.

The wing, which is similar to that used in other Ryan models is built in one continuous piece and has a cord of 7 ft. As in the case of other Ryan wings, the

The Wright 300 powered Ryan B-5 Brougham



A Ryan B-5 on Route

Clark "Y" aerial section is employed. The wing is of constant chord through out the span. Structurally, the wing consists of two spruce spars and 41 full width ribs gossiped with mahogany plywood. Drawn steel adjustable tie rods are used in the drag bracing and the ribs supporting the two 50 gal gasoline tanks are braced with steel compression members. The landing gear is covered with plywood and the wing tip section is faired with balsa wood. Grade A cotton flightex fabric through stretched by hand is used to cover the wing. Six coats of dope three chromum pigmented and sealed are then applied.

The fuselage structure is of welded steel tubing, 3025 S.A.E. specification being used except in the main members which are of chromum molybdenum. After the structure is assembled each section is subjected to an internal oil bath under pressure of 80 lb. per sq. in. The oil is then drained out leaving a thin protective coating which prevents external rusting. Rust preventives are also used on the outside of the tubing.

The cabin is entered through side type doors staggered for easy entrance and exit and the exterior finishing reflects the tendency in aircraft design to follow the contour work of naturalities. The cabin walls are insulated against temperature and sound and provided with large window space and easy opening windows equipped with curtains. A well designed heating system also is provided. A bathroom cabin scheme is also employed throughout and all seats are comfortably upholstered. The baggage compartments is accessible from the cabin.

An armrest fixture is found in the pilot's seat, which is located in a forward part of the cabin. This seat is adjustable sideways, backward and forward and, by a simple motion, its position can be changed to the center of the cabin from the left side to permit center control



Two photographs showing the method of entering the left seat when third on the right is not in use

convenient position on the instrument board.

All control cables and rods run beneath the flooring and none are visible in the cabin. Dual controls for each of the foot areas may be had or a center stick if the plane is not used for instruction purpose. The adjustable seatback is operated by a lever with five knotted positions in the pilot's left. Provision is made for ease of lubrication and inspection of all control wires and other moving parts. Water and air tight zipper fastenings enable the fabric to be open at several points

when the seat seat to it is not being used. The rig is built into a one-piece type which folds are truly out of the way and locks against the wall, also by a simple motion. Unusual views is afforded by the characteristic step-down shaped windows in the forward part of the cabin.

Complete instrument equipment is provided and includes compass, turn and bank indicator, tachometer, altimeter, oil pressure gauge, oil thermometer, oil level gauge, oil speed indicator.

The gasoline gauges on the wings are easily seen from the pilot seat. The compass is located on a spar over the pilot's head to remove it from any possible vibration from the ignition system. It is read through a mirror mounted in a

A specially designed exhaust heater is built into the exhaust duct, according to the manufacturer, a controllable temperature can be maintained in the cabin even during weather of ten deg. below zero. The heater is so designed that the force of the air traps and compresses air in a chamber around the exhaust. By this means a forced draft of heated air free from gasoline fumes enters the cabin through a floor grating.

Included in the power plant equipment of the plane is a new oil cooler which is installed in the tank located against the fire wall. The tank is hinged with small tubes to which a controllable supply of air runs. Other engine components are installed in all oil lines to eliminate possible breakage of the pipes due to vibration. The engine is also equipped with an air cleaner, gasoline strainer and standard luxury accessories. An exhaust pipe of the ducted or bypass type is employed. The plane is equipped with an adjustable Standard Steel

propeller. The engine is fitted with an engine fuel-injection system. A heavier magneto supplies a hot spark which insures easy starting in cold weather.

Landing gear is of the light axle type equipped with Aural shock absorber struts as standard equipment. The landing gear axles are of chrome molybdenum steel specially heat treated. Landing brakes are provided, each being operated independently by a slight pressure on the brake pedal located just beneath the rudder pedals. A tail wheel operating on ball bearings equipped with a pneumatic tire and a shock absorber of the rubber disk and steel spring type also is provided. Flight tests on the new plane revealed no position have recently been successfully completed under the supervision of inspection from the Department of Commerce. The weight empty with provision is 2850 lb.; useful load, 1530 lb.; pay load, 930 lb.; and gross weight, 4100 lb. The specifications are furnished to *AVIATION* by the manufacturer as follows:

Type	Basic monoplane
Size	42 ft. 4 in.
Chord	7 ft.
Length	28 ft. 3 in.
Height	9 ft. 9 in.
Weight empty	2850 lb.
Weight empty (portable)	2850 lb.
Useful load	1530 lb.
Useful load (portable)	1530 lb.
Pay load	930 lb.
Pay load (portable)	930 lb.
Gross weight	4100 lb.
Gross weight (portable)	4100 lb.
High speed	140 mph.
Cruising speed	120 mph.
Landing speed	50 mph.
Rate of climb	1200 ft. per min.
Service ceiling	18,000 ft.
Take-off run	275 ft.
Approved type certificate	No. 142
Engine	Wright Whirlwind J-6, 300 hp. at 2,000 r.p.m.



Upper the upper casing for examination of the ball wheel shock absorber mechanism. Below: Propeller air bearing structure for landing.



Ford Motor Company AND AMERICAN AERONAUTIC DEVELOPMENT

By JOHN T. NEVILL

THE SUMMER and fall of 1938 saw many progressive changes at Ford Airport. One of the most important of these was signaled by the test flights on the first tri-engine Ford monoplane to reach the test flying stage.

It will be recalled that the first tri-engine craft built by the company was destroyed in the fire which razed the airplane plant on the morning of Jan. 17 last year. The actual plane had been a Liberty-powered type converted into the tri-engine design by the installation of three Wright engines, and some changes in the cockpit. It was called the 3-AT.

Immediately following the fire the second three-engine plane was started in the company's temporary quarters. On June 11, long before the new and present factory building had been completed, this tri-engine plane known as the 4-AT, had been finished and test flown by Mr. R. W. Schneider, former Army pilot, who, at that time, was chief test pilot for the Ford company. This plane was powered by three Wright "Whirlwind" J-6 engines and was equipped to carry eight passengers. Having a wing span greater than that of the tri-engine Biplane in which Commander R. R. Byrd had reached the North Pole the preceding winter, it was the largest commercial airplane in America.

It will also be remembered that the Ford company had already announced plans for confining its production to the multi-engine class. Biplane had tests on the first ship been completed, therefore, before the second three-engine product was well under way.

The Ford's decision to produce only multi-engine craft for the market was actuated by the idea Ford's engineers held that the multi-engine class holds the greater degree of safety. His production schedule since that time has laid him up, unexpectedly, as the multi-engine idea of a problem still remains a moot question in the industry today. A single-engine, four-passenger all-metal experimental monoplane was built about a year later, but Mr. Ford never allowed it to be placed in production.

When this decision was put into effect, the company had built a total of eight Liberty-powered craft, according to a statement by William B. Mayo in Ford News of Aug. 7, 1936.

"Four of these," the statement read, "were sold to Florida Airways, Inc. (This sale was in December, 1935). The other four had been in service at the Ford

The Company Concentrates on Tri-Engine Planes; Ford Airport is Improved to Accommodate Lighter-Than-Air Craft; The "Air Flivvers"; Second Reliability Tour

airline. More planes were in process when the fire wiped out the factory.

"This fire left the Ford Motor Company with only four airplanes to operate two airlines. At once an attempt was made to see how far these planes could be forced, in a matter of hours per day in the air. Loads, originally 1,000 lb. of company freight, were gradually increased to 1,300 lb. or more.

"At first, one round trip to Cleveland and one to Chicago were made daily, but for the last few months the schedule in Cleveland has been increased to two round trips daily. This has been done with only three planes, the fourth being used in experiments in wing curve development. Thus the company has been running three airplane trips per day with only three planes."

"It is usual to have from four to six planes on the ground for each one in the air. In many cases eight crews are maintained to service the planes. We are, in process, making three round trips each day, operating only two planes—a condition which still exists. This means that the plane which flies to Cleveland and back in the morning, arriving home about 1 p.m., leaves for Chicago at 3 p.m. The plane which comes over from Chicago to the morning, arriving at noon, then off after lunch for a round trip to Cleveland. There is no sight crew for servicing these planes and we have two big hangar mechanics per engine in service. Our maintenance figures will undoubtedly surprise many in the industry. We are having them upon scheduled only performance and we now have an experience (Aug. 1, 1938) of 311,720

miles, or twelve and a half times around the world at the equator.

It is our belief that, with the development of motors to keep pace with plane construction we have now developed, twenty hours a day or more is feasible. It is evident that such mileage per day would require night flying.

"These designs made in 1926 for the Army were brought forth and various three-engine designs studied, with the idea of night flying in mind. Night flying, of course, makes use of the multi-power-plant idea, so that unobstructed unobstructed landings may be avoided. Also, the multi-engine ship permits much better vision than has been obtained on any previous single-engine large plane, and, though it requires more fuel capacity, it provides much greater cargo space."

Mr. Mayo also pointed out in his Ford News statement first, before offering Ford planes to the general public, the company had purposely tried first on the company's own airlines "to determine just what their capabilities were." This "demonstrating" time lasted over a period of approximately six months, the Ford airline opening April 13, 1925, the first plane being sold to John Wauwerman, Oct. 8, 1925, and four additional planes to Florida Airways, Inc., December, 1925. These five planes, he it is remembered, constitute all of the Liberty-powered aircraft built and sold by Ford. "There were, it is true, other companies operating Liberty-engined Ford monoplane, but such companies bought their planes from the original purchasers. A notable example was the State Air Services, Inc. This corporation, organized by William B. Stout as an operating company after he and his stockholders had sold their manufacturing plant to Ford, came into being Sept. 3, 1925. Its first plane was the Liberty-powered Ford 8 purchased from John Wauwerman. It then purchased two of the planes owned by Florida Airways. The fourth plane was a tri-engine

machine bought from Ford May 1, 1927. But that is another story.

During all of this period the Ford Motor Company was overlooking nothing that, in its belief, would improve the aircraft. This work, too various to describe in detail, was being carried on almost constantly. Recent developments, however, warrant the inclusion here of one particular project that got under way early in 1926. This was the Ford motorized car.

The great interest of both Henry and Edsel Ford in lighter-than-air craft is generally known. The Aircraft Development Corporation (now absorbed by the far more powerful and far more diversified Detroit Aircraft Corporation), and the Stout Metal Airplane Company, in fact, were in a way twin companies. The early Stout planes and the preliminary work on the present almost daily-built being built for the United States Navy, took shape under the same roof. And virtually the same group of men were interested essentially in both of the projects, among them Edsel B. Ford and William B. Mayo.

Shortly after completion of the morning mist in the spring of 1926, Henry Ford invited the Navy Department to fly the dirigible Shenandoah to Dearborn and make land. The invitation was accepted and the voyage started. That the trip was never completed is history. However, on Sept. 18, 1926, the Army dirigible R-5, commanded by Col. John A. Fagnolia, flew to Detroit from St. Louis, and was the first aircraft to use the Ford motor.

AMONG OTHER lighter-than-air facilities included at the A airport was the installation of piping and outlets for the filling of balloons, in the northeastern section of the field. A number of balloon events, including the start of two Gordon Bennett International balloon races, as well as many individual flights, have been made from the airport. Henry Ford was 62 years old on July 31, 1926. This



One of the early three-engine Ford transport planes, built by the Standard Oil Company.

occasion marked the revelation of another of the manufacturer's experiments—the much discussed "driver" plane. As in the case of the eight-cylinder airplane motor, Mr. Ford placed it in a group of newspapermen who were in Dearborn getting the usual "hardly any news." Until that time, the driver plane had received a name. True, for several weeks before it had been seen in test flights about the vicinity of the airport, but, as in the case of all Ford developments, verification of its existence could not be proved.

The Ford Motor Company has built and flown two such planes, both being designed and built for the proving in a small way of things that would have cost a great deal of money to do with a larger plane. At no time was any serious consideration given to building such a plane for the market. Both driver planes were of the single-plane, open, low-wing monoplane type and neither was of all-metal construction. The original had a 20-ft. wing span and weighed 120 lb. It was powered by a 25-hp, three-cylinder, Amundt air-cooled motor, and was said to have a high speed of 100 m.p.h. Some of its interesting features, besides its size and performance, include a pre-matto-steel tail wheel, a very wide wheelbase, and a muffler which threaded the exhaust nose to a remarkable degree.

Verifications indicated by the first driver plane justified construction of the second one. Although the second plane was slightly larger and heavier than the first, its general appearance was virtually the same. Its most interesting feature, no doubt, was its engine—a two-cylinder, 36-hp, air-cooled engine developed by Ford engineers.

This plane was not completed until late in 1927. Its engine weighed 118 lb. complete, of which 30 lb. was made up by its durable ignition system. The width of the motor was 15 in. and its length, 25 in. The cylinders were of cast-iron structure, the heads being made of high strength aluminum casting. Cylinder barrels were of steel, with the upper and threaded into the casting. Dow forged metal, one-fourth the weight of mild steel, but of equal strength, was employed in the pistons, connecting rods being of tubular wrought steel. The crank of the rod was built in two parts, hinged out, and electrically butt-welded together. This, it was claimed, afforded a weakness of strength with a reduction of weight, and insured something new in connecting rod

construction. The crankshaft had a roller bearing and a ball-bearing main bearing, which served as a thrust bearing as well as a medium through which oil was passed through the ball-bearing shaft to the connecting rod bearings. A gear case, attached to the front of the cast aluminum crankcase, enclosed the oil pump and valve gear, which operated the valves through enclosed push-rods. Oil was carried in an aluminum container bolted to the bottom of the crankcase. The engine was said to have been run continuously "for the block" for 50 hr.

BUT TO GET BACK TO 1926: Aug. 7, that year, saw the start of the second annual Ford airplane reliability tour for the Edsel B. Ford Reliability Trophy. Approximately twenty planes participated, the 2,884-mile route carrying the contestants across the states of Michigan, Illinois, Wisconsin, Minnesota, Iowa, Nebraska, Kansas, Kansas, Indiana, and Ohio. The tour was won by Walter Beach, flying an open cockpit Travel Air biplane, with the late Ernie Goldborough acting as navigator. Although the event proved highly pleasing from a performance standpoint, it was marred by several mishaps in which no one was injured seriously. Probably the most serious of these mishaps involved one of the two Ford planes participating in the tour. Maj. R. W. Schroeder, piloting a tri-engine machine (the first multi-engine craft built and flown by the company), suffered a forced landing near New York, due to a defective propeller on one of the wing engines. This induced such heavy vibration as to tear the engine loose from its frame, which, in turn, tore away a part of the craft's landing gear, and damaged the tip of the propeller on the opposite wing engine, making it necessary to cut the engine. Schroeder, however, skillfully brought the plane down without serious damage to the machine or injury to passengers. The type of propeller used was immediately discontinued and no further trouble of this sort has been experienced.

While the tour was in progress there occurred in Washington an event that has meant a great deal to American aviation. On Aug. 11, 1926, William P. MacCracken became Assistant Secretary of Commerce for Aeronautics.

The seventh of this series of articles by Mr. Mayall will appear in an early issue of AVIATION.



The Ford experimental plane used for World's "His America" R-7, which at the time was the world's fastest motor boat, on the Detroit River. The plane was.

Weight Control IN THE DESIGN OF AIRCRAFT

By FREDERIC FLADIER
Consultant Aircraft Corp.

THERE are two factors predominating in the performance of a heavier-than-air craft. These are its aerodynamic form and its weight. The designer of a successful airplane, which must be one of superior performance, is confronted primarily with the incorporation of aerodynamic excellence into the design and with the securing of the lightest structure possible, consistent with the necessary safety requirements.

It may be established that a unit of weight as an airplane is equivalent in its effect on any particular measure of performance, to a definite amount of resistance in terms of equivalent flat plate area. The equivalent of weight to resistance is variable with the speed of the aircraft and its attitude of operation. For high speed aircraft a larger increase of weight is permissible for a given reduction in resistance than is the case for a

Component	Weight (lb.)	Weight (lb.)	Weight (lb.)	Weight (lb.)
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200

Table 1 (continued)

Component	Weight (lb.)	Weight (lb.)	Weight (lb.)	Weight (lb.)
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200

Table 2

plane of lower speed. In the design of racing aircraft, the reduction of resistance is very important and a greater proportionate increase of weight may be expended to achieve this than in other types of aircraft. It appears then that in most designs not of the strictly high speed type, the reduction of weight may be more important than the attainment of resistance. Also, it shows the importance of weight increases. The extra resistance of an unswept engine at low altitudes outweighs the weight saving it makes possible. At high altitudes, however, this position is reversed. It is known that the performance of certain types of aircraft equipped with unswept engines is definitely superior to the same type provided with water cooled power plants when operated at high altitudes.

The designer must determine through performance calculations, the importance of weight as opposed to resistance for the particular aircraft engaging his attention. Having done so he must then consider the agency of having his airplane built according to the weight requirements laid down in the preliminary stages of the design. Lastly is the accurate prediction of the weights

of the component parts of the airplane in the design stage leads to very undesirable effects. An airplane may be designed to transport a definite useful load, and if, as frequently happens, the dead weight of the plane exceeds the value used in the stress analysis, a serious

Component	Weight (lb.)	Weight (lb.)	Weight (lb.)	Weight (lb.)
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200

Table 3 (continued)

reduction in the safety factor may occur. This may reduce the structural safety below the requirements which are definitely fixed by the civil or military authorities. If no reduction in the safety requirements is permissible, a reduction in useful load equal to the amount of overweight must be suffered. There are instances in which both civil and military planes of ordinary use have been built which have been 500 or more pounds overweight. A safety type designed for

a maximum load factor of 12 on the wings may actually be subjected to a load factor of 8 in a severe maneuver. If its design is based on a gross weight of 3,000 lb. and it is 500 lb. overweight, it may be subjected to a load factor of 8 based on its actual weight of 3,500 lb. The safety factor for a plane weighing 3,000 lb. would be only $12/8 = 1.5$, and for the plane which is 500 lb. overweight the safety factor would be roughly $12/8 \times 1.2 = 1.29$, a very serious reduction in safety. $12/8 \times 1.2 = 1.29$, a very serious reduction in safety.

Not only is the safety factor reduced by overweight but also the expected performance suffers correspondingly. In the case of a civil aircraft, the reduction in useful load in any considerable amount precludes the possibility of the airplane competing successfully with well engineered contemporaries.

The effective control of weight during the design and construction of an aircraft requires the constant application of rather meticulous care and attention to detail. It will usually be necessary to assign at least one person from an engineering department to the task, whose time will be entirely taken by it.

The method of weight control used by the Consolidated Aircraft Corporation in connection with the development of the Admiral PV-1 Navy Patrol Flying Boat involves the following procedure:

1. Estimated Weight Statement

A preliminary "weight estimate" of the dead weight should be made of all the component parts of the airplane by the designer. This should be done by the use of a standard form which properly lists all of the principal items of the airplane weight, together with a tabulation

Component	Weight (lb.)	Weight (lb.)	Weight (lb.)	Weight (lb.)
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200
Engine	1,000	1,000	1,000	1,000
Propeller	100	100	100	100
Wing	2,000	2,000	2,000	2,000
Fuselage	1,500	1,500	1,500	1,500
Landing gear	500	500	500	500
Control surfaces	300	300	300	300
Armament	200	200	200	200

4 Table showing the general method of weight comparison. (See drawings.)

THE Induced Drag VIEWPOINT

By CLARE B. MILLIKAN
California Institute of Technology

OF PERFORMANCE

THE subject of airplane performance from the point of view of the modern theory of induced drag has been very popular lately and has been treated in several recent papers.¹ The present communication does not discuss any results which are in principle new, but rather presents one of the standard formulas in a new form which is felt to have definite advantages, both in exhibiting the essential physical elements in performance, and also in making it easy for the designer to utilize the results. A certain amount of repetition of previous work is unavoidable, but this will be made as brief as possible.

The small assumptions are adopted, namely that the lift equals the weight of the plane and that the absolute coefficient of parasite drag is constant throughout the flying range. We include in the parasite drag both the so-called structural and also the wing profile drag. The lift, L , we assume is used solely otherwise specified. If it is the altitude the fundamental equation of performance may be written

$$\frac{dL}{dt} = 550 \frac{M_p - M_p}{W} \quad (1)$$

where dL/dt is the rate of climb in ft/sec, M_p is the thrust horsepower available, M_p is the thrust horsepower required, and W is the weight of the plane. Since the left side of (1) is a velocity it is convenient to introduce (following Schleich)²

$$w_0 = 550 M_p/W \quad (2)$$

w_0 is the familiar sinking speed and is the vertical speed of descent of the plane with power off, so represents the rate at which the plane would gain altitude if all of the thrust power available were utilized in lifting the plane vertically and may be called the "rising speed." Note that both are in ft/sec. Then (1) may be rewritten

$$dL/dt = w_0 - w \quad (3)$$

Formula for w_0 (Power Required)

From the definition of w_0 , we have

$$w_0 = \frac{DP}{W} = \frac{D_0 + D_1}{W} V \quad (4)$$

where D = total drag, D_0 = parasite drag, D_1 = induced drag (all as in previous), and V = velocity of flight in ft/sec. It is a well known result of the Prandtl wing theory that

$$D_1 = \frac{W^2}{\pi \rho V^2 (b)^2}$$

where ρ = density of the air and b = equivalent macro-

plane span.³ If we write F = "equivalent flat plate area" then

$$D_1 = 1.28 \frac{F}{V^2} = 0.64 \frac{F}{V^2}$$

Substituting these expressions into (4) we get

$$w_0 = 0.64 F \frac{1}{V^2} + \frac{1}{\pi \rho V^2} \frac{W^2}{(b)^2}$$

or, writing α = relative density of the air referred to sea level ($\alpha = 0.00283$), and defining

$$L_0 = \frac{W^2}{(b)^2} = \text{span loading}^4$$

$$L_1 = \frac{W^2}{F} = \text{parasite loading}$$

we arrive at the simple relation

$$w_0 = 0.08125 \frac{F}{V^2} + 25 F \frac{L_1}{W^2} \quad (5)$$

This expression is very interesting. Neglecting altitude effects for the moment, it shows that a given velocity the sinking speed (or M_p) at sea level depends on two aerodynamic parameters only, namely the span and parasite loadings. Span loading has not been a much used parameter in this country, but in Europe and especially England it is very often referred to in place of aspect ratio as a criterion to the comparison of planes. It has proven very useful in this connection and its physical importance as indicated by equation (4) shows that it marks a more general use in this country. With regard to the other parameter in (4), namely the parasite loading, it has most of the merits of span loading except that of general usage. It has been mentioned in one or two papers, but as far as the author is aware, has not as yet been used by the practical designer. It is obviously analogous to such more common parameters as "equivalent flat plate area," "parasite drag coefficient," and "parasite L/D ," but seems to have advantages over all of these quantities.

Introducing rough average values, say $L_1 = 2.5$ and $L_0 = 300$, into (4) the relative importance of the two parameters under various conditions is very clearly exhibited. Consider three average conditions: $V = 300$ ft/sec. (stalling speed), $V = 150$ ft/sec. (cruising

speed, speed for best climb), and $V = 200$ ft/sec (high speed). Then the expressions for w_0 are as follows:

$$\begin{aligned} w_0 &= 2.6 + 8.4 & (\text{stalling speed}) \\ w_0 &= 6.6 + 6.1 & (\text{speed for best climb}) \\ w_0 &= 10.6 + 3.4 & (\text{high speed}) \end{aligned}$$

where the first terms correspond to the first term of (5) and the second to the second term of (5). We have assumed sea level conditions, i.e., $\alpha = 1$. We see that at stalling speed the term involving span loading has over three times the importance of span loading; at cruising speed, at cruising speed or speed for best climb the two terms are roughly equal, while at high speed the parasite loading far outweighs the span loading term. The same relative importance of span and parasite loading for these three conditions will be found to hold approximately for all normal planes, and it is extremely important for the designer to have them well in mind. For long duration or altitude planes or for planes designed to fly economically at cruising speed he must pay attention to both terms, i.e., he must make the parasite loading as large as possible and the span loading as small as possible. For high speed planes, on the other hand, he may concentrate his attention primarily on making the parasite loading as large as he can.

Some interesting deductions as to altitude effects can also be drawn from (4). As altitude increases α decreases and hence at constant speed the first term in w_0 decreases while the second term increases. Hence the net result on w_0 depends on which of the two terms is the dominant one. At high speed w_0 (and hence M_p) decreases with altitude in constant speed. Near stalling speed w_0 increases as the altitude is increased, V remaining constant. If the speed is maintained near to the value for best climb at sea level and the altitude is increased w_0 , and M_p , remain nearly constant.

It is of some interest to have explicit expressions for w_0 effects for the moment, and this is attained getting expressions for the forward velocity V under these conditions. We shall briefly consider the two conditions of maximum overall L/D (denoted by the subscript L) and maximum sinking speed or power required (denoted by the subscript P). From the condition that L/D be a maximum it can easily be shown that $C_{D0} = C_{D1}$ and from the relation the following expression for the flying velocity at maximum L/D may be deduced. (The details of the analysis are not given since Deigl's paper previously mentioned contains them in a different form.)

$$V_L = \frac{100}{\sqrt{\alpha}} (L_0/L_1)^{1/4} \text{ (ft/sec.)} \quad (6)$$

or introducing the relative air density and expressing V_L in mph

$$V_L = \frac{10.0}{\sqrt{\alpha}} (L_0/L_1)^{1/4} = V_L \text{ for max. } L/D \text{ in mph} \quad (7)$$

Similarly for minimum power required the relation $C_{D0} = 3C_{D1}$ and from this we obtain for the velocity at which minimum power is required

$$V_P = \frac{0.738}{\sqrt{\alpha}} (L_0/L_1)^{1/4} \text{ (ft/sec.)} \quad (8)$$

or

$$V_P = \frac{10.6}{\sqrt{\alpha}} (L_0/L_1)^{1/4} = V_P \text{ for min power}$$

required in mph. (9)

The well known result appears that $V_P = 76 V_L$, i.e., the velocity for minimum power required is 0.76 times the velocity for maximum L/D , or the velocity for smallest sinking speed is 0.76 times the velocity for best gliding angle.

We can now substitute (2) and (5) into (4) and find the sinking speed for these two conditions. We get in this way

$$\begin{aligned} w_0 &= \frac{25}{\sqrt{\alpha}} (L_0/L_1)^{1/4} = \text{sinking speed} \\ &\quad \text{at max. } L/D \text{ (ft/sec.)} \\ w_0 &= \frac{32.9}{\sqrt{\alpha}} (L_0/L_1)^{1/4} = \text{minimum sinking speed} \\ &\quad \text{(ft/sec.)} \end{aligned} \quad (10)$$

The relation between the two quantities is

$$w_{0P} = 0.88 w_{0L}$$

From (7) the horsepower required for a particular plane at the two conditions can of course be calculated using (2).

Considerations Involving w_0 (Power Available)

Our considerations so far have been concerned entirely with the power required part of performance. For any performance prediction or analysis for power driven planes the above results must be combined with power available data. The most desirable thing would of course be to develop an analytical expression for w_0 as a function of V , and this is in fact done in (4). Unfortunately it has been impossible as yet to find any such accurate analytical formula which is general and yet simple enough to be of practical service. The number of variables involved is too large, including both engine and propeller characteristics, and at least in the case of those associated with the latter, the available information is essentially empirical and tabular in nature rather than theoretical and analytic. This is the reason why the power available portion of any accurate performance estimation is almost invariably tabular or graphical in nature, and this in turn makes it practically inevitable that the power required portion of the analysis be expressed in the same form. This would appear to detract considerably from the importance of an analytic expression such as (4), at least in routine aircraft performance estimation. However, even for such work it is probably at least as simple to plot a curve for w_0 or M_p from equation (4) as to plot the curve from a table of computed values in the normal manner. There is also some advantage in using curves of w_0 and M_p in V , instead of the customary M_p and M_p vs. V , since in this way the effect of change of excess horsepower is read directly from the chart.

It should be mentioned in this connection that several attempts have been made to develop an analytical expression for w_0 . Schleich in particular (loc. cit.) has considered the question in detail and has developed several interesting approximate expressions. However, in view of the difficulties mentioned above, the practical usefulness of these formulas seems doubtful, at least until engine data is available to a more general and analytical form than at present.

Level High Speed

In view of the difficulties encountered above in constructing M_p and M_p vs. V to get reliable expressions for complete performance, there is one instance in which such a construction can be effected satisfactorily. This is the case of full speed level flight, for at this condition the brake horsepower available is assumed to be the rated full load power of the engine vs. M_p and the

¹ W. C. Schleich, paper, *A. S. E. Monthly News*, June 1958, pp. 10-12; also *Aviation*, August 1958, pp. 10-12.

² The quantity w_0 is the velocity V at which the plane would sink if all of the thrust power available were utilized in lifting the plane vertically and may be called the "rising speed." Note that both are in ft/sec. Then (1) may be rewritten

³ In particular, W. Schleich, *NACA Tech. Mem.* No. 471, and *Aviation*, August 1958, pp. 10-12.

and propeller characteristics can vary rapidly and L_0 far as existing plane from the observed high speed, and can thus use this value in the estimation of L_0 for a new and similar machine.

Fig. 1 should also assist in determining the propeller efficiency η (assuming propeller interference effects to be neglected). Knowing the design R.P.M. and the propeller diameter, the design maximum efficiency can be determined from the well known curve of maximum efficiency vs. design $P/(ND^5)$ of P is known. Assuming a value of P/η is determined and then from Fig. 1 the actual P/η can be read off. Using this P/η a new value of η is found and the process repeated until the value of P assumed in the determination of η and the value obtained from Fig. 1 for this η agree. This can be carried out.

For comparison purposes of very different existing aircraft it is better to make the comparison using the loading factor W/S as a parameter in the location of the aerobically equivalent of parasite drag and coefficients resulting in the existing drag factor C_{D0} .

TECHNICAL REVIEWS

N.A.C.A. Report No. 216—Tables for pressure of air on coming to rest from Various Speeds, by A. F. Zah and F. A. Louie.

In Technical Report No. 216 of the N.A.C.A., theoretical formulas are given from which was computed a table for the pressure of air on coming to rest from various speeds, such as those of aircraft and propeller blades. In this report, the table is accompanied and additional static pressures of air for air-speed intervals in miles per hour and for some cross-speed intervals in knots per hour, kilometers per hour, and meters per second. The pressure values in Table II are also more exact than the values given in the previous table used to furnish the aerodynamic engineer with ready essential formulas for finding the pressure of air on coming to rest, Table I has been derived for the standard values specified below. This table first presents the theoretical pressure-speed intervals and their working formulas for several special units of speed.

N.A.C.A. Technical Note No. 512—The Use of Wheel Brakes on Airplanes, by Thomas Carroll and Smith I. D'Avanzo.

This report discusses the use of wheel brakes upon airplanes. The results of tests to determine the effect of wheel brakes on the landing run of an airplane under various conditions of load and in various wind velocities are presented. The advantages of the use of brakes in reducing the landing run and in increasing the facilities of ground maneuvering are discussed, together with methods of operation and application.

N.A.C.A. Technical Note No. 387, Strength of Tubing Under Combined Axial and Transverse Loading, by L. S. Twissman, S. S. Petrovich and C. D. Johnson, Bureau of Standards.

It is believed by the Materials Committee that this report contains the first values for combined loading on tubes which are based on experimental data. Curves and tables given in the report are said to be sufficiently accurate for use in design of a tubular member which

must vary very rapidly until η has been satisfactorily determined.

Certain of the standard formulas relating to power required have been given in a form which brings out the nature of the physical dependence of the phenomena on two aerodynamic parameters; span loading and parasite loading. The implications of these formulas have been discussed in some detail. The difficulties in getting accurate analytical expression for power available in general have been recognized and have been shown to disappear in large measure for the case of high speed level flight. A simple analytical formula for maximum speed has been obtained which has also been exhibited in graphical form. This formula involves the two most important aerodynamic parameters and an advance over previous quantities, the thrust power loading. The possible uses of the high speed relation have been briefly considered.

acts both as a column and a beam. Tests of duralumin tubes are in progress and will be reported later.

N.A.C.A. Technical Note No. 510—Wind Tunnel Pressure Distribution Tests on a Series of Biplane Wing Models, Part I. Effects of Changes in Stagger and Gap, by Montgomery Knight and Richard W. Noyes.

This report furnishes information on the changes in the factors on each wing of a biplane, called when other the stagger or the gap is varied. The data were obtained from pressure distribution tests made in the Ames-Whipple Wind Tunnel of the Langley Memorial Laboratory. Since each test was carried up to 50 deg angle of attack, the results may be used in the study of stalled flight and spinning as well as in the structural design of biplane wings.

N.A.C.A. Technical Note No. 306, Pressure Distribution on a Model R.A.F. 21 Airfoil in the Variable Density Wind Tunnel, by Herbert N. Jacobs.

Excerpt—Kodak Company booklet X-ray in Industry: A brief survey of the theory and use of X-rays in industry.

S.A.E. Semi-Annual Meeting paper, Multicamber Distribution in Multi-Cylinder Engines, by M. J. Zarewsky.

S.A.E. Semi-Annual Meeting paper, Report on Air-Fuel Ratio Tests, by H. W. Best, Sheffield Scientific School, Yale University.

Written discussion of M. W. Best's paper, by Charles Louis Fritz, Jr., Aeronautical Engineering Staff, Massachusetts Institute of Technology.

S.A.E. Semi-Annual Meeting paper, Effects of a Centrifugal Supercharger on Fuel Vaporization, by C. F. Fritz Taylor.

S.A.E. Semi-Annual Meeting paper, Combustion Chamber Designs in Theory and Practice by W. A. Whitcomb.

American Society for Testing Materials paper, Abrasive Alloys for Pressure Die Castings, by Sam Toy.

A.S.M.E. Aeronautics Division Paper—Aircraft for Passenger Transport, by Charles N. Moultrie.

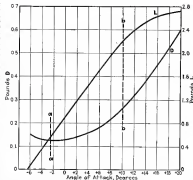
THE Penaud VERSUS THE Canard TYPES

By ALBERT MERKILL

THE Penaud type is here defined as a monoplane having spaced apart fore and aft, the front surface or surfaces being the supporting means, the rear surface acting as a stabilizer. The Canard type is a system of surfaces spaced apart fore and aft, the rear surface or surfaces being the supporting means, the front surface acting as the stabilizer. The Penaud was a Frenchman who did a lot of valuable aerodynamic research work in the twenties of the last century. The Canard type is so named from its resemblance to a duck with its head protruding in front and its tail

If the supporting means consists of one surface both of these types are called monoplanes whereas in fact both are biplanes, naturally as two surfaces are necessary for pitch stability. Moreover, since the surfaces are spaced apart fore and aft and (if the c.g. is properly placed) for static pitch stability there must be downlift to obtain the proper trim angle. It follows that every so-called monoplane is, in fact, a staggered-downlift biplane. If normal flying range is defined as the range of angles covering the angle for maximum speed and the steepest and cruising angles, then for flying throughout the range there is an necessity of going outside this range except when one has to dodge a solid obstacle, when aerodynamic help may be necessary. Also as long as one flies within this range the airplane can never get out of control in pitch, yaw or roll, and there can be no nose over, stall or spin.

The above graph shows the variation of lift and drag in pounds with angle of attack, D being the total resistance to the flight. The normal flying range lies roughly between the two lines ab and bc , differing, of course, with every model. Let us compare the Penaud with the Canard type on a basis of class examples. The Penaud has the stabilizer at the rear, while the Canard has it at the front, both types being steered to rapid bodies in that comparison. Note that within the normal flying range the rate of change of L with angle is greater than is the rate of change of D with angle; that is the curve



of L is steeper. This means that within the normal flying range L dominates over D in determining the sign and magnitude of the static pitching moment couple. It follows therefore that within this range, stable static pitching moment characteristics can be made the same for both types by placing the c.g. properly and having the proper amount of downlift. In this range of angles the Canard has an advantage inasmuch as its stabilizer lifts, whereas the Forward stabilizer carries a down load.

If now we consider the condition in the right of the line ab we get a very different result. Here we find that the rate of change of L with angle of attack is greater than is the rate of change of D , and this means that D will dominate. It is just this that will always make the Canard type very dangerous in my opinion. In the Penaud type the D of the stabilizer is always a lifting force, like the feathers on an arrow, but in the Canard the D is always an upsetting force because the stabilizer is in front and at those angles where D dominates over L this upsetting force is going to cause accidents.

British Methods of STEEL AIRCRAFT CONSTRUCTION

By W. H. SAVERS

IN THE preceding article an account was given of the methods of steel construction employed by two aircraft manufacturers in Great Britain. The work of the remaining firms which have contributed to the development of this type of construction will be outlined in this, the concluding article of the series.

Boislin & Paul Ltd., Norwich

As with the two firms previously considered, Boislin & Paul Ltd. have been engaged in the development of corrugated strip construction since before the end of the War. At the Salon D'Aeronautique held in Paris in 1933 Boislin & Paul exhibited the P10, which was as far as is conceivable the first complete airplane constructed throughout from strip steel. This machine is shown in Fig. 29. The fuselage was framed in steel with longtruss and end frames of drawn sections, and the wings had spars of box section of corrugated strip.

Except for the covering which was of fabric for wings, etc., and of a special Dacron material for the fuselage, the whole of the P10 was of steel.

Since that time this firm has developed a mixed type of construction wherein steel is used for the more heavily loaded members and the light alloys in members less heavily stressed. There are several marked advantages in this system. One of the most important from the practical point of view is that it is possible to make the same section in, say, steel or in duralumin, and to obtain in either material practically the same strength weight ratio. For example, when two struts of the same overall length, one in every less than the load of the other, are required, they may be made of the same dimensions and form, one in steel, the other in duralumin. This materially reduces the cost of tools, etc.

Moreover, there are many cases where steel sections for members such as ribs must be made unnecessarily strong owing to the difficulty of handling the very thin gages required, or where sections thick enough to stand up to normal flying stresses would be too delicate and flimsy to stand handling on the ground at the hands of the usual type of mechanic. In these cases the thicker parts called for by the normal stress requirements will give in the light alloy member that robustness which may be lacking in the normally equally strong steel section.

This use of mixed materials is to be found throughout the structure of the firm's products. It extends even to the construction of fittings, which though not actually essential parts of the structure yet require some greater degree of robustness than can be provided by fabric and formers, yet do not justify the use of steel covering. Such items are made of spruce and duralum. By this choice of the material which may be most economically

employed in each specific component remarkably low component weights are a realized feature.

Wing spars are generally of the box type. Fig. 30 shows two examples of Boislin & Paul spars, one dating from 1919, the other a present-day production which is used on the "Sedgeland" twin engine bomber. Both sections have corrugated flanges, webs with fairly wide central fins which later are supported and stiffened by a number of tubular distance pieces of special form. The only difference of note between these two spars are that as the more recent type the crown of the flanges has been raised further from the level of the joints at the edges, thus improving the effectiveness of this member and decreasing the stress intensity at the edges of web and flange, and that light cross strips have been added across these joints, thus ensuring the maintenance of accuracy of width across the joint line. As these lips are used to locate ribs this dimensional accuracy is a great aid to accurate assembly.

A further difference not revealed by the figure is that the earlier spar was made from strip drawn in the hardened state whereas the modern sections are formed from annealed strip and are hardened and tempered after forming.

Fig. 31 shows a range of standardized spars made by this firm. It will be seen that there are on this figure three horizontal rows, each of six sections. All are sections in any one vertical row share the same web section. Each spar in any one of the six vertical rows has the same flange section. Thus a total of nine component sections, making for nine sets of spars, produces eighteen spar sections. As each of the nine basic sections can be produced in several gages of steel on the



Fig. 29

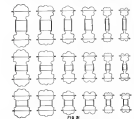
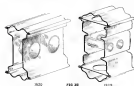
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same tools, or if desired, duralumin may be substituted for steel, an enormous range of complete spars is available from a strictly limited tool equipment. The range actually available is suitable for almost any size of machine between the limits of 2,000 to 30,000 lb. loaded weight.

Fittings for struts, etc. are attached to these spars by the use of side plates riveted to the outer sets of flanges



on each web. As the webs are standardized so also are the dimensions of these reinforcing plates.

Assembly of such spars is by no means the expensive process that it might appear. The two web members are first assembled with their intervening distance tubes, which are span down on the outside of the webs. The flanges with the stiffener strips already mentioned are then drawn over the lips of the webs and the component is then placed in a semi-automatic machine which spars

and drills the rivet holes in all four points simultaneously. While one set of holes is being drilled by the machine, rivets are being inserted in the last set of holes and clenched by a hand operated single point by the machine assistant. In the event of a need for very great increased production, the development of an attachment which will automatically insert and clamp the rivets would present no difficulty. The labor cost of producing a spar of this type is not as low as that of making a normal duralumin or alloy spar of equal size and strength and, owing to the limited number of tools required under this system, the cost of tools is very moderate.

For internal drag struts, and for fuselage longtruss and struts, tubes of circular section are used. Where such tubular members are very heavily stressed and consequently tubes of normal pipe in diameter also may economically be used, solid drawn circular tubes hardened and tempered are used. For example, the central section of a fuselage and the engine mountings might be made of such tubes. For conditions where an external section would be of an irregularity this pipe is a solid drawn tube the firm produce the "lock-joint" tube section shown in Fig. 32. This is made from strip material on the draw-bench in either steel or light alloy in lengths of up to 75 ft., perfectly straight and to such dimensional accuracy that machined sockets and struts will fit into it to give certain interchangeability. Tubes of this type made in a stainless steel with an ultimate tensile strength of over 90 tons per sq. in. are used for all the main longitudinal girders on R.101, the new 5,000,000 cu ft. airship now being completed at the Royal Airship Works, and several miles of such tubes were made by Boislin & Paul to which many thousands of tapered fittings were attached without any need for hand fitting.

Such accuracy of dimension is not easily obtainable in any class of solid drawn tubing and certainly not in such control of such high tensile. Even more important than the accuracy of external dimensions of these tubes is their almost perfect uniformity of wall thickness.

The range of sizes in which this type of tube is made ranges from 14 to 24 in. diameter with wall thicknesses of as little as .010 in. in the largest sizes. Practically any desired steel may be used as well as duralumin or a similar light alloy. The steel usually used in this form is a nickel-chrome-nickel alloy, formed soft and hardened and tempered to give 75-80 tons per sq. in. tensile strength.

Intertruss and other struts which are exposed to the wind are of composite construction, consisting of an equal section metal nose, and a light rail fitting of three-ply with spruce formers. As shown in Fig. 33 the nose portion is made of a leading trough section in which is riveted a rear channel. The timber forming strips into the





push-in on the rib, and provides four external air faces in two pairs mutually at right angles. Two cylinders, slightly staggered one to the other, are pressed through both sleeve and lugs. These serve the same purpose as the wing plates and provide pin-joint attachments for socket ends joined to the struts.

Fig. 35 shows the standard sleeve section, and Fig. 36 a typical fuselage joint. Sheets of this type are also applicable to other types of joint as, for example, that shown in Fig. 37.

This shows a joint for the attachment of a wing spar to the fuselage of a high performance twin-engine bomber. A machined link, pinned to the fuselage cross-member, has welded to it a pair of heavy flange plates. The ends of two sections of the fuselage lugs are welded to these flange plates and are bolted to them. One of these ends that for the spar section, is provided with a sleeve of the standard section to provide the necessary joint force. The other lugs are welded, which has to be made considerably heavier, but a special steel is used. The standard joint sleeve section is also used at the end of the vertical fuselage strut at this point.

To transmit the large shear loads which may pass from spar to other fuselage to the thin welded tubes used in this type of construction without either relying on the relatively weak sleeve material or using additional



large diameter pins or bolts through the tubes "thru-bolts" are employed. These are short flanged hardened steel bolts just long enough to pass through the walls of sleeve and tube at each side, and drilled to take the full shear and torque in the case of joints such as those of Fig. 37. When a socket to transmit tension only is to be utilized in a thin welded tube, "thru-bolts" held in place by tubular covers may be used.

The use of a variety of metals in the same aircraft

structure as in this firm's practice renders it more than usually important to insure adequate protection against corrosion, and the several materials which are so well known only have adopted their such research on this aspect of metal construction.

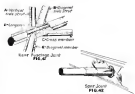
Of the various materials used, duralumin is that most apt to succumb to the action of corrosion. British experience has shown that the process known as "anodic oxidation" carried out under proper control, and with due regard to the composition of the particular alloy, followed by a coat of lacquer gives very satisfactory results if no conditions likely seriously to aggravate corrosion are allowed to occur. But it is extremely important in a metal structure to prevent corrosion of any metal other than the duralumin itself from occurring in contact with that material.

Booth & Paul has succeeded in producing the necessary types of metal sections for their own type of construction in stainless steel, as has already been noted. But this material is expensive and difficult to work. The nickel-chromium-nickel steel which they normally employ for formed sections is much easier to work and far more and one in its quality than the standard varieties at present available. Moreover, it has a very close and uniform metal structure, and is able to take to corrosion



attack under normal conditions. Properly cleaned, and coated with a suitable lacquer, it is practically corrosion proof.

Wiring plates and similar details are made from stainless steel, and machined parts, including screw threads, are also made by a zinc vapor process which permits the deposition of an adherent, non-porous coating, whose



thickness can be kept to within sufficiently few limits not to interfere with interlocking ability. Sheets for duralumin tube members—or steel ones where the loading permits—use the coating of a heat-treated aluminum alloy which has excellent corrosion resisting qualities.

Machines of this make have now been in service for several years without signs of trouble from corrosion. One reason for the opportunity arose to make a performance test on a machine of this construction after two years of hard service. The result of this test was practically identical with that obtained on its original test, and rather better than the reverse. This fact should go to substantiate the claim that a metal treated machine will maintain its condition in service better than will a timber framed one, and to dispense the suspicion that the British type of corrugated strip construction is as fragile as it looks to eyes unaccustomed to it.

The Bristol Aeroplane Co. Ltd.

It has already been recorded that the first British machine embodying the corrugated type of strip construction was designed and built by this firm as long ago as 1918, and so this it may now be added that as the result of a competition held by the British Air Ministry late in 1927 the "all-metal" "Bristol" design was chosen by the same firm as adopted as the standard type for the re-equipment of the fighting squadrons of the Royal Air Force.

This machine, which is shown in Fig. 38 is fitted with a Bristol "Tiger" supercharged engine developing 420 hp at 12,000 ft. Empty the machine weighs 1,960 lb. It carries 685 lb of fuel and oil and 528 lb of military load including pilot, gun, ammunition, weapons and receiving system, and engine, making a total weight of 3,125 lb.

At speed, at 13,120 ft., the following performances were attained:

Time from Start	Height	Level Speed m.p.h.
30	5,000	177
60	10,000	174
95	15,000	168
142	20,000	162
205	25,000	162
325	30,000	162

These figures themselves are sufficient evidence that machines of this construction are not in any way handicapped in regard to weight and performance by the

methods employed to produce them. In this connection it is worthy of note that the makers guarantee to reject the performance specified to within plus or minus 5 per cent on any machine of this type.

The Bristol company uses steel exclusively for structural members of their all metal airplanes. For a machine with normal wing bracing arrangements box type spars of corrugated section are employed. The web and flange forms are generally similar to those used by the last firm dealt with. Some steel spars for cantilever wings have also been produced by this firm which employ corrugated flange sections with lattice type shear bracing. Some examples of deep tapered spars of this type are shown in Fig. 39, and Fig. 40 shows an assortment of strength comparisons of their ribs.

An interesting item in this figure is the section of fuselage frame which is entirely made from laminated strip. Longitudinal and struts are made from two strips each rolled or drawn to an arc of a circle with two projecting lips. The two lips are riveted together to form a T-type tube. The longitudinal have ribs at 90 and the struts at 180 deg. and joints between the two types of members are made by means of plates riveted to the lip of both members as shown in Fig. 41.

Solid drawn tube is used in this firm's practice for interplane struts, drag struts and the forward heavily stressed section of fuselages. In the case of the "tail-drag" above mentioned, the interplane struts and wing bracing wires are attached not to the spars directly but to steel tube drag struts which pass through stiffener plates riveted to the sides of the spar and through the spar itself. This arrangement is indicated in Fig. 42.

The H. G. Hawker Engineering Co. Ltd.

This firm is a comparatively less eminent into the ranks of steel construction. Its efforts have been largely directed by the development of methods of construction of the simplest possible type. For wing spars they have developed the "double right" section shown in Fig. 43



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This is produced by rolling an initially round solid drawn tube to shape, and is obviously a cheap and simple spar to produce. It is only possible to produce this section in steel of medium tensile strength and is relatively thick gauge and it is therefore heavier than the strip type so far described. Expressed as a percentage of spar weight this excess weight is quite considerable, but in steel machines the net additional to the total weight of the machine is relatively small and this type of spar has obvious merits. However, it is not possible to neglect



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even relatively small increases of weight in airplane structures if they can be avoided and the firm has recently developed the type of spar shown in Fig. 44. This is of the strip type very similar in form to the Armstrong Whitworth type, but has tubular booms formed from one width of strip and has only two rows of struts, thus making a step toward cutting out this process of joining the components of this type of spar.

Ribs by this firm are in general of aluminum or of duralumin, stamped from sheet metal, though a variety of types have been tried and still more are, it is believed, in course of trial.

The method of fuselage construction developed by this firm is of very considerable interest on account of its simplicity. Circular section solid drawn tubes are used for bulk longitudinal and struts. Where joints occur the longitudinal tubes are flattened by a special hand rolling machine and flat fish plates are attached to these flats by one cup headed bolt. Struts in the plane of the fish-plates are provided with squared or dished ends which fit between the plates and are fixed by spun hollow nuts. Struts in the plane at right angles are provided with ball-bush which fit into the cup-heads of the bolts mentioned. The side members of the fuselage are made as separate rails with rigid tubular lattice bracing. Top and bottom rails have ball-headed struts and wire cross-bracing. Fig. 45 shows a typical joint of this type.

Conclusion

This article is by no means a complete account of all British achievements in steel construction. Such a complete description would occupy an inordinate amount of space. For this reason it is possible that many details which deserve attention have been omitted or too briefly mentioned. It is desired to express to all the firms mentioned the writer's thanks for the information which they supplied so free and without which this article could not have been written.

In conclusion it should be understood that most of the types and methods of construction described are fully covered by patents held by one or another of the firms who have been mentioned.

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"B" went after it differently. He appropriated \$125,000 for general consumer mediums and \$25,000 for the papers of his trade. He intended to create distribution direct and along with consumer demand. He recognized that if the dealer had a place in his sales program then the dealer's magazine had a place in his advertising program. *Radio Retailing* was used as his key paper.

A recent accounting shows that "B" sold three times as many radio sets as "A" during 1928.

This is one of a series of advertisements devoted entirely to advertising in the radio field. It is intended to show the reader that the advertiser who is not a radio advertiser is not a radio advertiser. It is intended to show the reader that the advertiser who is not a radio advertiser is not a radio advertiser.

There are some advertising men like "A" who currently believe that consumer demand by itself automatically organizes distribution. Over a long period it may possibly do so. But what about the manufacturer who cannot afford the long-haul expense—the manufacturer who must seek his profits from economies in marketing as well as production economies—the manufacturer bent with serious price competition?

The wrong use, the improper balance of advertising has often turned users of advertising into foes or skeptics. It is not a question of one type of medium in preference to another but a matter of common-sense principles of merchandising. With these understood and applied an effective, economical program of advertising is inevitable.

McGraw-Hill Publications

New York Chicago Cleveland Detroit Philadelphia St. Louis
Cincinnati San Francisco Boston London

Hawks and Wasp

break more trans-continental records



Captain Frank M. HAWKS

New York to Los Angeles in 19 hours, 39 minutes, 23 seconds. The old figure was 24 hours, 51 minutes.

The new non-stop time from Los Angeles to New York is 17 hours, 38 minutes, 36 seconds. Captain Hawks and the "Wasp" held the former record of 18 hours, 21 minutes.

The elapsed time from New York's Roosevelt Field to Los Angeles and back again was 44 hours, 48 seconds... actual flying time 36 hours, 46 minutes, 48 seconds... also a record. It was the first time in history that man had flown coast to coast and return in such remarkable time. To set these new marks across the skyways of the continent, the "Wasp" traveled approximately 5500 miles at an average speed of 144 miles per hour. The outstanding performance characteristics of the "Wasp" are thus again strikingly emphasized by conspicuous accomplishment.

THE
PRATT & WHITNEY AIRCRAFT CO.
HARTFORD - CONNECTICUT
Division of United Aircraft Corporation

Manufactured in Canada by The Canadian Pratt & Whitney Aircraft Co., Ltd., Longueville, Quebec, in Continental Europe by the French Pratt & Whitney, Marly.

Wasp & Hornet

Engines

KITTY HAWK

Flies Like a Hawk—Lands Like a Kitten

A Kinner Powered Ship—Unusually Economical



The new Model B-1 Junior Kitty Hawk is a two-seater, high-wing, conventional landing gear, light engine, easy to fly, economical, reliable, and rugged. It is a true "Kitty Hawk" in every sense of the word.

WILLIAM AERONAUTS, INC.
BOSTON, MASS.

TITANINE

Registered Trade Mark

A Complete Range of AIRPLANE FINISHING MATERIALS

Dope—proof paint, black, white, aluminum or grey.
Clear varnish and varnish dopes.
Ti-Two colored non-corrosive dopes.
Fingerprinted nitrate dopes.
Flexible wing lacquer.
Wood and metal lacquer (for inside and outside fields).

TITANINE, Inc.
Union, Union County, New Jersey
Distributors to the United States Government

The Budd Wheel Company

Makers of
Wheels for airplanes

PHILADELPHIA AND DETROIT

For AVIATION INSURANCE SEE AUER

WE HAVE THE LOWEST RATES
AND THE BEST SERVICE
AND THE MOST COMPLETE
AND MOST COMPLETE
AND MOST COMPLETE
AND MOST COMPLETE

Don't Overlook Opportunities

Men who regularly keep in touch with the market through other channels often overlook the many opportunities that are to be found in the

SEARCHLIGHT SECTION
For Every Business Want
"The SEARCHLIGHT First"



PROPELLERS HARTZELL

20 Years Experience Behind Our Service

HARTZELL PROPELLER CO. PIQUA, O.



FAIRCHILD cabin mono-planes and open biplanes form a complete line of quality aircraft for dealers' sales. Fairchild aviation products also include pistons, oils, landing lights and engines. Dealers address: Fairchild Airplane Manufacturing Corporation, Farmingdale, L. I., U. S. A.

AIRPLANES



NICHOLAS-BEAZLEY... IS AS NEAR YOU AS YOUR TELEPHONE
CALL MARSHALL, MO. 1400

"Same Day" Shipment on Supplies You Need... Complete Aeronautical Stocks at Your Command!

Shock Cord —1/2" x 10'—\$1.00 1/4" x 10'—\$1.00 1/8" x 10'—\$1.00	Landline —1/2" x 10'—\$1.00 1/4" x 10'—\$1.00 1/8" x 10'—\$1.00	Flightline —1/2" x 10'—\$1.00 1/4" x 10'—\$1.00 1/8" x 10'—\$1.00	Flying Tools Screwdriver—\$1.00 Wrench—\$1.00 Pliers—\$1.00 Saw—\$1.00 Hammer—\$1.00 Nails—\$1.00 Screws—\$1.00 Washers—\$1.00 Rivets—\$1.00 Bolts—\$1.00 Nuts—\$1.00 Lock Washers—\$1.00 Lock Nuts—\$1.00 Lock Bolts—\$1.00 Lock Nuts—\$1.00 Lock Bolts—\$1.00
Aviation Lights Day—\$1.00 Night—\$1.00 Day—\$1.00 Night—\$1.00	Tires and Tubes 1/2" x 10'—\$1.00 1/4" x 10'—\$1.00 1/8" x 10'—\$1.00	American Propellers 1/2" x 10'—\$1.00 1/4" x 10'—\$1.00 1/8" x 10'—\$1.00	WATER BOMBS 1/2" x 10'—\$1.00 1/4" x 10'—\$1.00 1/8" x 10'—\$1.00

Nicholas-Beazley
Airplane Company Inc.
MARSHALL, MO.

NOTRUS HANGAR CORP.

Standard Sizes For Department of Commerce Class "A"—"B"—"C" Airports

Ready, quickly, and economically moved—designed under federal approval as the field means for dismantling and moving—Sturdy and convenient with substantial doors—large side space for office, shop or storage—Removes the hazard of wind-blown debris, and provides for easy access—Non-breakable color-glass windows.

Save a third. Write for bulletins.

NO-TRUS HANGAR CORP.
Main Office: Dayton, OH, U.S.A.
Eastern Office: 140 Cedar St., New York, N. Y.



JAEGER

2 Day Work Chronograph 3 Day

Time of Flight Watch, 3 Day

For complete information, please write to the Jaeger Watch Co., 100 N. 1st St., Philadelphia, Pa. 19106. We will send you a complete catalog of our watches and chronographs, and a list of our dealers and agents in the United States and abroad. We will also send you a list of our dealers and agents in the United States and abroad.

AERONAUTICAL TIMEPIECES





Series of photos shows (clockwise) some air engineering work in this

Airports and Aviation Buildings by Austin

COMPLETE Airport Service—professional surveys and reports, engineering surveys and studies, site selection.

Design and Construction—graveling, grading, lighting, all necessary airport buildings such as hangars, depots, etc. Also aircraft factory buildings.

Ask for appointment, costs and booklet "Airports and Aviation Buildings."

THE AUSTIN COMPANY
Airport Engineers and Builders

Cleveland
New York, Chicago, Philadelphia, Detroit, Cincinnati, Portland,
St. Louis, Seattle, Cincinnati, Los Angeles, San Francisco, Dallas



BOYCE MOTO METER

The MotoMeter Company and its subsidiary, the National Gauge & Equipment Company, offer a complete line of instruments best indicators, oil pressure and oil pressure gauges.

Long experience in the construction of these instruments, coupled with large production, makes it possible to offer high grade instruments at reasonable prices. Every meter has been taken to insure the dependability of these instruments.

Best Indicator
Pressure Gauges



Oil Pressure Gauge
and Meter

Oil Pressure Gauge
and Meter

Oil Pressure Gauge
and Meter

MotoMeter Gauge and Equipment Corporation
2 Wilbur Avenue, Long Island City, N. Y.



LIGHT WEIGHT

SPECIALLY designed for airplanes, the Bendix Wheel and Bendix 2-Stroke Servo Brake form a single unit—sturdy, efficient, compact.

The 30x5 inch size weighs 22½ pounds complete with brake.

Now in production in standard sizes.

Circle protected by patents and specifications in U.S. and abroad

BENDIX BRAKE COMPANY

General Offices and Plant South Bend, Ind.

Division of Bendix Aviation Corporation, Chicago

BENDIX ④ BRAKES

FOR SAFETY

WE MAKE

on our automobiles, thousands of parts of alloy steel, hardened and heat-treated to meet the most exacting service. Do you AVIATION PEOPLE need products of such quality?

If so—let's get together.

STANDARD PRESSED STEEL CO.

Jenkintown, Pa., Box 328



THE FLIGHT MARKER



And records last of your trip airplane by recording altitude, speed, direction, etc. The instrument is so simple that the operator can use it without special training. It is a valuable record of your trip. See our new product.

EPC AIRCRAFT UTILITY, 100 South St., College Point, L. I.

WEAKS SEC for machinery AVIATION



Mr. Gordon, who is shown wearing a Gordon Aerotog, says, "The quality and design of the Gordon Aerotogs certainly make them a valuable addition to my flying wardrobe."

Mr. Gordon is a qualified pilot and has flown over 100,000 miles in the 1938 Gordon Aerotog. Most of his flying has been in the Gordon Aerotog.

There is a reason why Gordon Aerotogs are so popular. They are made of the finest materials and are built to last.

Mr. Gordon is a qualified pilot and has flown over 100,000 miles in the 1938 Gordon Aerotog. Most of his flying has been in the Gordon Aerotog.

EXPERIENCED



The J. H. B.

BERLIN-JOYCE

AIRCRAFT CORPORATION

BALTIMORE, MARYLAND

World A. Berlin, Chief of Materials, a former chief manufacturing and, more recently, general engineering consultant on specifications for the military, aviation, boat construction and aviation. The J. H. B. is a former factory representative of the Berlin-Joyce Aircraft Corporation at Philadelphia.

COLONIAL GRAIN UPHOLSTERY LEATHER

made by

EAGLE-OTTAWA LEATHER COMPANY
Grand Haven, Mich.

Genuine leather for airplanes is NON-INFLAMMABLE durable and clean.

Simple facts furnished on application

Circle 16
1000 Washington Blvd.
St. Louis, Mo.
1000 Lumber St.
St. Louis, Mo.
1000 Lumber St.
St. Louis, Mo.

"Opportunity" Advertising: Think "SEARCHLIGHT" First!

UNIVERSAL
AVIATION CORPORATION
LANSBURY, ST. LOUIS AIRPORT, ANGLIM, MISSOURI

THANK YOU for advertising AVIATION

ALUMINUM AND ITS ALLOYS for Aircraft

Aluminum Company of America

2482 Oliver Bldg.

Pittsburgh, Pa.

1400 Airplane Motors R-10 and R-11 AIR-COOLED

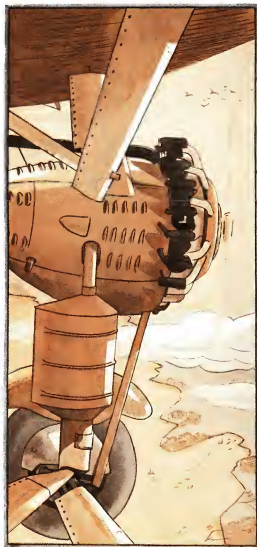
Simple and successful plans for converting to
FIXED RADIAL
R-10 and R-11 R.P.

See Mr. Smith at Booth No. 222
Cleveland Show, Aug. 24th to Sept. 2nd
Cleveland Hotel at night

Permanent Address: TIPS & SMITH, Inc.

P. O. Box 103, Warren, Ohio

THE NAME OF WRIGHT IS ONE OF AVIATION'S ASSETS



MILLIONS of men have yet to fly. Some will never do so. But as succeeding thousands come to decide that they will ride the air the name of Wright is one of Aviation's assets.

For in the minds of millions the name of Wright... the reliability of Wright... the endurance of Wright... give confidence and courage to the neophyte of the air when for the first time, he sweeps upward in the skies.

The name of Wright is one of aviation's assets. It is so not to the lone benefit of ourselves, but to every man who builds a plane—to every man who takes his living from the air. For the job of all of us in aviation is to build this kind of confidence; to work together as business men should; to take from aviation its heroic tinge, and to clothe it with confidence and every-day reliability.

Last year set a new record in the total of passengers carried for hire. 1929 will see more millions climb into planes for their first trips. Many more will enter flying schools. Many will buy their own planes. All are hastened in these decisions by their knowledge of Wright's proven dependability.



WRIGHT
AERONAUTICAL CORPORATION
PATERSON, NEW JERSEY

